

**NASA TECHNICAL NOTE**

NASA TN D-5910



**NASA TN D-5910**

C. I

LOAN COPY: RETURN  
AFWL (WLOL)  
KIRTLAND AFB, NM

0132745



TECH LIBRARY KAFB, NM

**A METHOD FOR THE THERMAL ANALYSIS  
OF SPACECRAFT, INCLUDING ALL MULTIPLE  
REFLECTIONS AND SHADING AMONG  
DIFFUSE, GRAY SURFACES**

*by Doyle P. Swofford*

*Langley Research Center  
Hampton, Va. 23365*

**NATIONAL AERONAUTICS AND SPACE ADMINISTRATION • WASHINGTON, D. C. • JULY 1970**



|   |  |  |                             |
|---|--|--|-----------------------------|
| 1. Report No.<br>NASA TN D-5910   | 2. Government Accession No.                                    | 3. Recipient's Catalog No.<br><b>0132745</b> |                             |
| 4. Title and Subtitle<br><b>A METHOD FOR THE THERMAL ANALYSIS OF SPACECRAFT,<br/>INCLUDING ALL MULTIPLE REFLECTIONS AND SHADING<br/>AMONG DIFFUSE, GRAY SURFACES</b>  |  | 5. Report Date<br>July 1970                  |                             |
| 7. Author(s)<br>Doyle P. Swofford   | 6. Performing Organization Code                                |  |                             |
| 9. Performing Organization Name and Address<br>NASA Langley Research Center<br>Hampton, Va. 23365   | 8. Performing Organization Report No.<br><b>L-6063</b>         |  |                             |
| 12. Sponsoring Agency Name and Address<br>National Aeronautics and Space Administration<br>Washington, D.C. 20546   | 10. Work Unit No.<br><b>124-09-24-03</b>                       |  |                             |
| 15. Supplementary Notes   | 11. Contract or Grant No.                                      |  |                             |
| 16. Abstract<br><p>A new method which uses finite surface elements has been developed and used to calculate temperature histories for spacecraft of arbitrary physical geometry. This method assumes gray surfaces with diffuse reflection and radiation properties and accounts for all multiple reflections. The analysis is performed in terms of the Cartesian coordinates of the four corners of each plane quadrilateral element. Shading, or optical blocking, is accounted for in the radiant heat exchange between each pair of surfaces and in the thermal fluxes from external sources which are incident on each surface. This method has been programed for a digital computer and the program can be used as is for the thermal analysis of any spacecraft whose surfaces may be approximated by planar quadrilateral elements which obey Lambert's cosine law. A listing of the program and its auxiliary programs is included in the report. An example problem of a complex, multiwinged earth-orbiting satellite is also presented.</p> | 13. Type of Report and Period Covered<br><b>Technical Note</b> |  |                             |
| 17. Key Words (Suggested by Author(s))<br>Spacecraft thermal analysis<br>Multiple diffuse reflections<br>Digital computer programs  | 18. Distribution Statement<br><b>Unclassified - Unlimited</b>  |  |                             |
| 19. Security Classif. (of this report)<br><b>Unclassified</b>   | 20. Security Classif. (of this page)<br><b>Unclassified</b>    | 21. No. of Pages<br><b>63</b>                | 22. Price*<br><b>\$3.00</b> |

**A METHOD FOR THE THERMAL ANALYSIS OF  
SPACECRAFT, INCLUDING ALL MULTIPLE REFLECTIONS AND  
SHADING AMONG DIFFUSE, GRAY SURFACES**

By Doyle P. Swofford  
Langley Research Center

**SUMMARY**

A new method which uses finite surface elements has been developed and used to calculate temperature histories for spacecraft of arbitrary physical geometry. This method assumes gray surfaces with diffuse reflection and radiation properties and accounts for all multiple reflections. The analysis is performed in terms of the Cartesian coordinates of the four corners of each plane quadrilateral element. Shading, or optical blocking, is accounted for in the radiant heat exchange between each pair of surfaces and in the thermal fluxes from external sources which are incident on each surface.

This method has been programmed for a digital computer and the program can be used as is for the thermal analysis of any spacecraft whose surfaces may be approximated by planar quadrilateral elements which obey Lambert's cosine law. A listing of the program and its auxiliary programs is included in the report.

An example problem of a complex, multiwinged earth-orbiting satellite is also presented.

**INTRODUCTION**

For a spacecraft which has extended members with large surface areas or has large enclosed spaces containing personnel or sensitive electronic equipment, shading and multiple reflections among its surfaces can be critical to its thermal design.

The projected areas of the external surfaces of the spacecraft and the shape factors between all surfaces show the effect of shading of spacecraft surfaces by one another. The usual method of obtaining projected areas is by photographs of a model of the spacecraft. Shape factors may also be obtained by a photographic method (ref. 1, pp. 399-402).

For complex spacecraft in which temperatures are required at literally hundreds of nodal points, computer calculation methods are the only practical approach because of the large number of shape factors and projected areas that must be obtained.

When the computer program in this report was written, other programs were already available which calculate projected areas and shape factors for configurations with shading and which account for multiple diffuse reflections of thermal radiation as well. (See ref. 2.) However, since these programs require much time to prepare and run, a more simple, yet flexible program, such as the one in the present investigation, was desired for development use in a wide variety of spacecraft. The thermal design may be fairly well fixed by use of the simpler program and then confirmed by use of a more sophisticated program.

The treatment of the multiple reflections is made tractable by the usual approximation that all the surfaces reflect and emit radiation diffusely (i.e., according to Lambert's cosine law). Once the shape factors and projected areas are calculated, the quadrature solution to this diffuse problem (developed by Gebhart, ref. 3) can be used. The basis of the approach used here is to divide the irregular quadrilateral surfaces of the spacecraft into grids by the use of two-dimensional coordinate systems embedded in each separate surface. The division allows the required integrations to be carried out in two-dimensional spaces. A shading test for each grid element is made by determining whether a line from the source point to the grid point is intercepted by any other surface belonging to the spacecraft.

Calculated temperatures resulting from the shape factors and projected areas can be steady state or time dependent. If radiant heat transfer is predominant over conductive heat transfer, equilibrium temperatures may be found by solving simultaneous equations for the fourth power of the temperature of each surface. The nonequilibrium case (varying external and internal heat loads) is solved by integrating the time derivatives of the temperatures. In this case heat conduction is included in the calculation.

## SYMBOLS

A area

[A] diagonal matrix of the areas of a group of surfaces

$A_{\text{proj}}$  projected area of a surface

[a] diagonal matrix of surface absorptivities

B thermal flux incident on one surface from others

{B} one-dimensional array of thermal flux incident on each surface from the other surfaces

|           |  |
|-----------|--|
| c         | specific heat of a material  |
| $D_{ij}$  | radiation coefficient between two identically shaped, closely spaced parallel plates, $\frac{\sigma}{\frac{1}{e_i} + \frac{1}{e_j} - 1}$ |
| [D]       | matrix of the radiation coefficients $D_{ij}$  |
| e         | hemispherical emissivity of a surface  |
| [e]       | diagonal matrix of the emissivities of a group of surfaces   |
| $F_{ij}$  | shape factor between surface $i$ and surface $j$   |
| [F]       | square matrix of shape factors between pairs of a group of surfaces  |
| $f_{a,i}$ | shape factor of a spacecraft surface $i$ for the albedo flux from a planet surface   |
| $f_{p,i}$ | shape factor of a spacecraft surface $i$ for planet-emitted radiation  |
| [G]       | square matrix which yields $\{B\}$ when it operates upon $\{\epsilon\}$ (see eq. (6))  |
| H         | sum of the thermal flux emitted by a surface and the flux reflected by it  |
| $\{H\}$   | one-dimensional array of total thermal flux moving away from each surface  |
| $h_{ij}$  | conduction coefficient between nodes $i$ and $j$   |
| [h]       | square matrix of conduction coefficients between pairs of a group of nodes   |
| [I]       | identity matrix  |
| $I_{sh}$  | shading indicator ( $I_{sh} = 0$ : shaded by an intervening member; $I_{sh} = 1$ : not shaded)   |
| K         | sunlight indicator ( $K = 1$ : sunlit; $K = 0$ : shaded by planet)   |
| m         | mass   |

|                         |  |
|-------------------------|--|
| $\tilde{n}$             | unit normal vector of a surface  |
| $q$                     | net heat flow rate to a node   |
| $q_{\text{cond}}$       | net rate of heat flow to a node by conduction  |
| $q_{\text{int}}$        | rate of internal heat generation in a node   |
| $q_{\text{rad}}$        | net rate of heat flow between close parallel nodes by radiation                                    |
| $\bar{r}$               | position vector of a point   |
| $\bar{r}_{ij}$          | position vector of point $j$ relative to point $i$ , $\bar{r}_j - \bar{r}_i$                       |
| $S$                     | solar flux at position of planet or spacecraft   |
| $T$                     | absolute temperature, $^{\circ}\text{K}$   |
| $T_p$                   | equivalent blackbody temperature of a planet, $^{\circ}\text{K}$                                   |
| $t$                     | time   |
| $\bar{v}_c$             | position vector of the centroid of an elemental area   |
| $\bar{v}_{j,k}$         | position vector of the lower left corner of the $jk$ th grid element of a plane quadrilateral      |
| $\bar{v}_p$             | unit vector directed toward the center of a planet from an orbiting spacecraft                     |
| $[W]$                   | diagonal matrix with diagonal $W_i = \sum_j D_{ij}$  |
| $z_i$                   | coefficient for heat conduction away from a node, $z_i = \sum_j h_{ij}$                            |
| $[z]$                   | diagonal matrix of coefficients for heat conduction away from each node                            |
| $\alpha, \beta, \gamma$ | parameters which each give the ratio of the lengths of two colinear vectors having a common origin |

|                        |  |
|------------------------|--|
| $\alpha', \beta'$      | abscissa and ordinate of a point in a normalized skewed coordinate system  |
| $\epsilon$             | sum of the thermal emission flux from a spacecraft surface and the reflected portion of incident thermal flux arriving at the surface directly from sources external to the spacecraft |
| $\{\epsilon\}$         | one-dimensional array of the fluxes $\epsilon$ of the surfaces of a spacecraft   |
| $\theta_i$             | angle formed by vectors $\bar{n}_i$ and $\bar{r}_{ij}$   |
| $\theta_j$             | angle formed by vectors $\bar{n}_j$ and $\bar{r}_{ij}$   |
| $\theta_s$             | angle formed at the center of a planet by lines to the sun and to an orbiting spacecraft   |
| $\lambda_1, \lambda_2$ | parameters giving any vector lying in a given plane as a linear combination of two given vectors in the plane  |
| $\mu$                  | ratio of the projected area of a surface to its total area (unit projected area)   |
| $\{\mu\}$              | one-dimensional array of unit projected areas  |
| $\rho$                 | reflectivity of a surface  |
| $[\rho]$               | diagonal matrix of reflectivities of surfaces  |
| $\rho_p$               | reflectivity, or albedo, of a planet surface for solar radiation   |
| $\sigma$               | Stefan-Boltzmann constant  |
| $\Phi_{ji}$            | total radiative heat flow from surface $j$ which is incident directly upon surface $i$   |
| $d\phi_{ji}$           | heat flux incident upon elemental area $dA_i$ which is received from $dA_j$  |
| $\phi_a$               | planet solar albedo flux incident upon a surface of a spacecraft   |
| $\phi_p$               | planet thermal emission incident on a spacecraft surface   |

$\phi_{\text{rad}}$  net thermal radiation flux received by a plate  $i$  from identical plates  $j$  parallel to and near it,  $D_{ji}(T_j^4 - T_i^4)$

$\phi_s$  solar radiation flux incident upon a spacecraft surface

$\psi_{ji}$  function which is integrated to yield  $F_{ji} = \frac{(\bar{n}_i \cdot \bar{r}_{ij})(\bar{n}_j \cdot \bar{r}_{ij})}{\pi(\bar{r}_{ij} \cdot \bar{r}_{ij})^2}$

$\omega$  half the angle subtended by a planet from the position of an orbiting spacecraft

Subscripts:

$a$  albedo

$p$  planetary

$s$  solar

$t$  thermal radiation emitted by spacecraft surfaces or planet surface

When an expression with an asterisk affixed as a superscript has a negative value, it is set equal to zero; positive values are unchanged.

A bar over a symbol indicates a vector quantity.

## ANALYSIS

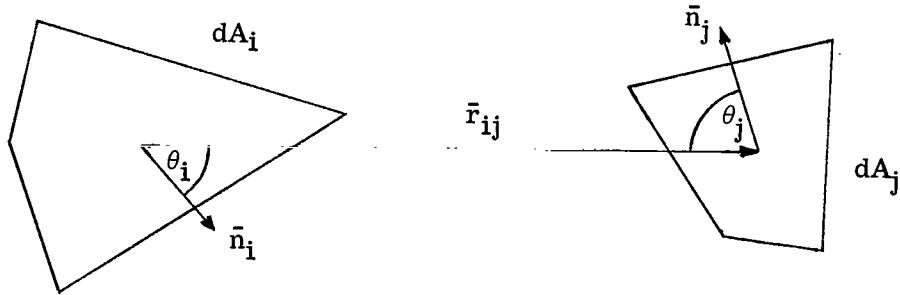
### Radiative Heat Balance

The analysis of the heat transfer by diffuse emission and reflection of thermal radiation between one surface and the remainder of a group of surfaces is introduced by first considering just two surfaces  $i$  and  $j$  exchanging thermal radiation. The two surfaces are assumed to be isothermal.

The heat flux incident upon the elemental area  $dA_i$  of surface  $i$  which is received from  $dA_j$  of surface  $j$  is given by

$$d\phi_{ji} = \frac{H_j}{\pi} \frac{dA_j}{r_{ij}^2} \cos \theta_i \cos \theta_j$$

where  $H_j$  is the heat flux leaving  $dA_j$ ,  $r_{ij}$  is the distance between  $dA_i$  and  $dA_j$ , and  $\theta_i$  and  $\theta_j$  are the angles made with the normals to  $dA_i$  and  $dA_j$ , respectively, by the line  $\bar{r}_{ij}$  between them as shown in sketch A:



Sketch A

If  $\bar{r}_i$  is the position vector of  $dA_i$ , and  $\bar{r}_j$  that of  $dA_j$ , then

$$\cos \theta_i = \bar{n}_i \cdot \frac{\bar{r}_j - \bar{r}_i}{|\bar{r}_j - \bar{r}_i|} = \frac{\bar{n}_i \cdot \bar{r}_{ij}}{|\bar{r}_{ij}|}$$

and

$$\cos \theta_j = \bar{n}_j \cdot \frac{\bar{r}_i - \bar{r}_j}{|\bar{r}_i - \bar{r}_j|} = \bar{n}_j \cdot \frac{-(\bar{r}_j - \bar{r}_i)}{|\bar{r}_{ij}|} = -\frac{\bar{n}_j \cdot \bar{r}_{ij}}{|\bar{r}_{ij}|}$$

where  $\bar{n}_i$  and  $\bar{n}_j$  are unit normal vectors to  $dA_i$  and  $dA_j$ . Thus,

$$d\phi_{ji} = -\frac{H_j (\bar{n}_i \cdot \bar{r}_{ij})(\bar{n}_j \cdot \bar{r}_{ij})}{\pi r_{ij}^4} dA_j$$

Now, let

$$-\frac{(\bar{n}_i \cdot \bar{r}_{ij})(\bar{n}_j \cdot \bar{r}_{ij})}{\pi r_{ij}^4} = \psi_{ji} = \psi_{ij}$$

so that

$$d\phi_{ji} = H_j \psi_{ij} dA_j$$

If  $\Phi_{ji}$  is the total heat flow incident upon surface  $i$  from surface  $j$ , then

$$d^2\Phi_{ji} = d\phi_{ji} dA_i = H_j \psi_{ij} dA_i dA_j$$

and

$$\Phi_{ji} = \int_{A_j} H_j \int_{A_i} \psi_{ij} dA_i dA_j$$

Assuming  $H_j$  constant over surface  $j$  gives

$$\Phi_{ji} = H_j \int_{A_j} \int_{A_i} \psi_{ij} dA_i dA_j = H_j F_{ij} A_i A_j$$

where the shape factor  $F_{ij}$  is the average value of  $\psi_{ij}$  over the two surfaces. Unlike the conventional shape factor,  $F_{ij}$  is not nondimensional and must be multiplied by  $A_j$  to yield the conventional quantity  $f_{ij}$ . Also,  $F_{ji} = F_{ij}$ . The average flux incident on surface  $i$  from surface  $j$  is defined as

$$B_i = \frac{\Phi_{ji}}{A_i} = F_{ij} A_j H_j$$

This is the flux incident on surface  $i$  from a single surface  $j$ . For more than one surface  $j$ , the total flux incident on  $i$  is given by the sum over  $j$  of the individual contributions:

$$B_i = \sum_j F_{ij} A_j H_j \quad (1)$$

If surface  $i$  is concave,  $B_i$  includes a contribution from surface  $i$  to itself. When  $B_i$  and  $H_i$  are treated as the  $i$ th components of linear arrays  $\{B\}$  and  $\{H\}$ , it follows directly from equation (1) that

$$\{B\} = [F][A]\{H\} \quad (2)$$

where  $[F]$  is a square matrix with elements  $F_{ij}$ , and  $[A]$  is a diagonal matrix whose diagonal  $A_i$  is the area of surface  $i$ .

The flux  $H_j$  leaving surface  $j$  is made up of  $\epsilon_j$  (the emitted flux and the reflected portion of flux incident from sources external to the spacecraft) and the reflected part  $\rho_j B_j$  of the incident flux from other surfaces, where  $\rho_j$  is the reflectivity of surface  $j$ . Thus,

$$H_j = \epsilon_j + \rho_j B_j \quad (3)$$

and  $\epsilon_j$  includes the reflected part of any external radiation directly incident on  $j$ , such as sunlight. In the array form,

$$\{H\} = \{\epsilon\} + [\rho]\{B\} \quad (4)$$

where  $[\rho]$  is a diagonal matrix of the reflectivities of the surfaces.

Equations (2) and (4) are two matrix equations in the two arrays  $\{B\}$  and  $\{H\}$ ; therefore, unique solutions may be obtained for them. Substituting equation (4) into equation (2) gives

$$\{B\} = [F][A]\{\epsilon\} + [F][A][\rho]\{B\} \quad (5)$$

and, on solving for  $\{B\}$ ,

$$\{B\} = [I - FA\rho]^{-1}[FA]\{\epsilon\} = [G]\{\epsilon\} \quad (6)$$

where  $[I]$  is an identity matrix. Since  $[G]$  is a function only of the surface properties and geometry, it need only be evaluated once for each configuration.

The net heat fluxes are given by the difference  $\{B\} - \{H\}$  between the incident flux and the flux leaving the surfaces. From equation (4),

$$\{B\} - \{H\} = [I - \rho]\{B\} - \{\epsilon\}$$

or

$$\{B - H\} = [a]\{B\} - \{\epsilon\} \quad (7)$$

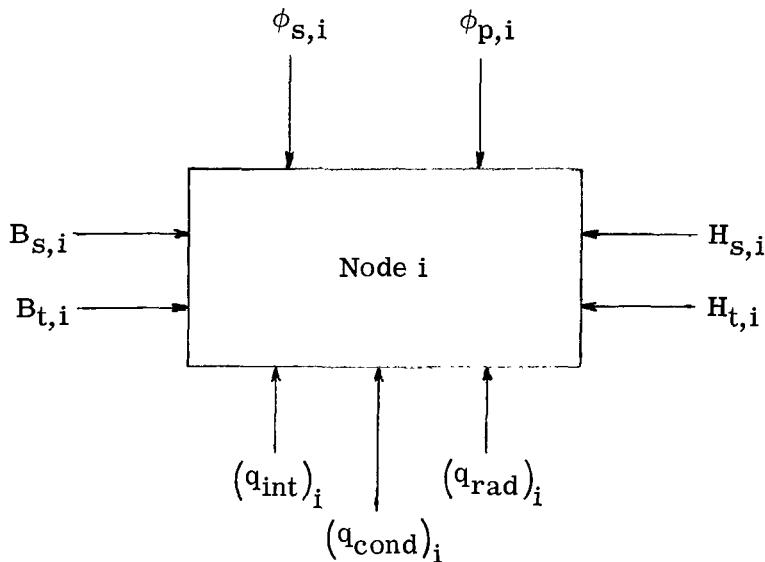
When equation (7) is substituted into equation (6), the following equation is obtained:

$$\{B - H\} = [aG - I]\{\epsilon\} \quad (8)$$

where  $[a]$  is a diagonal matrix of the absorptivities of the surfaces and the matrix  $[G]$  is given in equation (6).

Equation (8) gives the net flux on each surface due to all emissions and reflections of thermal radiation from all the surfaces. Now, only the fluxes from external radiation sources which are directly incident on the surfaces remain to be accounted for.

In the specific case of a spacecraft immersed in a real environment, there will be two regimes of thermal radiation flux – the principally short-wavelength solar flux and the principally long-wavelength emission from the spacecraft surfaces and from the planet surface. The external thermal radiation sources are direct and planet-reflected (albedo) solar flux and emission from the surface of a planet due to its temperature. Heat conduction as well as radiation will be accounted for. A diagram of the heat transfer for a single node is shown in sketch B:



Sketch B

If  $q_i$  is the net rate of heat input to member  $i$ , then the heat balance is given by

$$\begin{aligned} \frac{q_i}{A_i} &= \frac{m_i c_i}{A_i} \frac{dT_i}{dt} = (B_s - H_s)_i + (B_t - H_t)_i + (\phi_s + \phi_p)_i \\ &+ \frac{(q_{int})_i + (q_{cond})_i}{A_i} + (\phi_{rad})_i \end{aligned} \quad (9)$$

where

$m_i$  mass of  $i$

$c_i$  specific heat of  $i$

$T_i$  absolute temperature of  $i$

$\phi_{s,i}$  solar flux directly incident on  $i$ , both planet reflected and direct from the sun

$\phi_{p,i}$  flux emitted from the planet surface and directly incident on  $i$

$(\phi_{rad})_i$  net radiation flux to  $i$  from  $j$ , which is parallel to and near  $i$

$(q_{int})_i$  rate of internal heat generation

$(q_{cond})_i$  net rate of conduction to  $i$  from other members

Equation (8) gives for the solar spectrum

$$\{B_s - H_s\} = [a_s G_s - I]\{\epsilon_s\} \quad (10)$$

where  $\epsilon_{s,i}$  is the reflected part of the incident solar and albedo flux on  $i$ , that is,

$(\rho_s \phi_s)_i$ . For long-wavelength flux

$$\{B_t - H_t\} = [e G_t - I]\{\epsilon_t\} \quad (11)$$

where  $\epsilon_{t,i}$  is made up of reflected planet-emitted flux and the thermal emission of  $i$ .

Thus

$$\epsilon_{t,i} = \sigma e_i T_i^4 + \rho_{t,i} \phi_{p,i}$$

or in array form,

$$\{\epsilon_t\} = \sigma [e] \{T^4\} + [I - e] \{\phi_p\} \quad (12)$$

where  $\sigma$  is the Stefan-Boltzmann constant,  $[e]$  is a diagonal matrix of the emissivities of the surfaces, and  $\{T^4\}$  is an array of the fourth powers of the temperatures.

Equation (9) becomes, in the array form,

$$\left\{ \frac{q}{A} \right\} = \{B_s - H_s\} + \{B_t - H_t\} + \{\phi_s + \phi_p\} + \left\{ \frac{q_{int}}{A} \right\} + \left\{ \frac{q_{cond}}{A} \right\} + \{\phi_{rad}\} \quad (13)$$

Substituting equations (10), (11), and (12) into equation (13) gives

$$\begin{aligned} \left\{ \frac{q}{A} \right\} = & [a_s G_s - I][\rho_s] \{\phi_s\} + \sigma [e G_t - I][e] \{T^4\} + [e G_t - I][I - e] \{\phi_p\} + \{\phi_s + \phi_p\} \\ & + \left\{ \frac{q_{int}}{A} \right\} + \left\{ \frac{q_{cond}}{A} \right\} + \{\phi_{rad}\} \end{aligned} \quad (14)$$

and after simplification,

$$\begin{aligned} \left\{ \frac{q}{A} \right\} = & [a_s][I + G_s \rho_s] \{\phi_s\} + [e][I + G_t(I - e)] \{\phi_p\} - \sigma[e][I - G_t e] \{T^4\} \\ & + \left\{ \frac{q_{int} + q_{cond}}{A} \right\} + \{\phi_{rad}\} \end{aligned} \quad (15)$$

The net rate of heat conduction to member  $i$  is given by

$$(q_{cond})_i = \sum_j h_{ij} (T_j - T_i) = \sum_j (h_{ij} T_j) - \left( \sum_j h_{ij} \right) T_i$$

where  $h_{ij}$  is the conduction coefficient between nodes  $i$  and  $j$ . In array form,

$$\{q_{cond}\} = [h]\{T\} - [z]\{T\} = [h - z]\{T\} \quad (16)$$

where  $z$  is a diagonal matrix with  $z_i = \sum_j h_{ij}$ .

The net radiant heat to  $i$  from the identical panel  $j$  with which it forms an isolated enclosure is given by

$$(\phi_{rad})_i = \sum_j D_{ji} (T_j^4 - T_i^4)$$

where  $D_{ji}$  is the radiation coefficient between  $j$  and  $i$ , given by

$$D_{ji} = \frac{\sigma}{\frac{1}{e_i} + \frac{1}{e_j} - 1} = D_{ij}$$

Thus,

$$\{\phi_{rad}\} = [D - W]\{T^4\} \quad (17)$$

where  $W$  is a diagonal matrix with  $W_i = \sum_j D_{ij}$ .

Substituting equations (16) and (17) into equation (15) gives the heat balance as

$$\begin{aligned} \left\{ \frac{q}{A} \right\} = & [a_s][I + G_s \rho_s]\{\phi_s + \phi_a\} + [e][I + G_t(I - e)]\{\phi_p\} \\ & - \sigma[e][I - G_t e]\{T^4\} + [D - W]\{T^4\} \\ & + \left[ \frac{1}{A} \right] [h - z]\{T\} + \left\{ \frac{q_{int}}{A} \right\} \end{aligned} \quad (18)$$

Only the external sources  $\phi_{s,i}$ ,  $\phi_{a,i}$ , and  $\phi_{p,i}$  remain to be evaluated. The solar flux  $\phi_{s,i}$  incident on  $i$  is the simplest, being proportional to the projection of the sunlit area of surface  $i$  upon a plane perpendicular to a line to the sun  $(A_{proj})_i$ :

$$A_i \phi_{s,i} = (A_{proj})_i K S$$

$$\phi_{s,i} = \left( \frac{A_{proj}}{A} \right)_i K S = \mu_{s,i} K S$$

Or, in array form,

$$\{\phi_s\} = K S \{\mu_s\} \quad (19)$$

where  $K$  has the value 0 if the spacecraft is in the shadow of the planet and the value 1 if not; and  $S$  is the local solar flux.

The planet-emitted flux is given by

$$\phi_{p,i} \approx f_{p,i} \sigma T_p^4$$

and, in array form,

$$\{\phi_p\} = \sigma T_p^4 \{f_p\} \quad (20)$$

where  $f_{p,i}$  is the shape factor of surface  $i$  for the planet-emitted radiation, and  $T_p$  is the equivalent blackbody temperature of the planet ( $\sigma T_p^4$  is equal to the average value over the planet of the emitted thermal radiation flux).

The planet solar albedo flux on surface  $i$  is given by

$$\phi_{a,i} \approx \rho_p S f_{a,i} \cos^* \theta_s$$

or, in array form,

$$\{\phi_a\} = \rho_p S \cos^* \theta_s \{f_a\} \quad (21)$$

where  $\rho_p$  is the mean reflectivity of the planet for sunlight,  $f_{a,i}$  is the shape factor of surface  $i$  for the planet albedo flux,  $\theta_s$  is the angle formed at the center of the planet by the lines to the spacecraft and the sun, and  $\cos^* \theta_s = \cos \theta_s$  if  $\cos \theta_s$  is positive and equal to 0 if  $\cos \theta_s$  is negative.

The factors  $f_p$  and  $f_a$  are difficult to calculate, even without the added complication of partial shading of a surface. An approximation for them – exact when the panel is exposed to the entire portion of the planet surface which is visible from the position of the spacecraft – is

$$f_{a,i} \approx f_{p,i} \approx (\bar{n}_i \cdot \bar{V}_p)^* \sin^2 \omega \approx \mu_{p,i} \sin^2 \omega$$

Here  $\bar{V}_p$  is a unit vector pointing from the spacecraft toward the center of the planet, and  $\omega$  is the angle included by  $\bar{V}_p$  and a tangent from the spacecraft to the planet surface. The factor  $\mu_{p,i}$  approximates the effect of shading on  $f_{p,i}$  and  $f_{a,i}$ .

When the relations for the external fluxes (eqs. (19) to (21)) are substituted into equation (18), the heat balance becomes

$$\begin{aligned} \left\{ \frac{q}{A} \right\} &= KS[a_s][I + G_s \rho_s]\{\mu_s\} + \rho_p S \cos^* \theta_s \sin^2 \omega [a_s][I + G_s \rho_s]\{\mu_p\} \\ &+ \sigma T_p^4 \sin^2 \omega [e][I + G_t(I - e)]\{\mu_p\} - \sigma [e][I - G_t e]\{T^4\} \\ &+ [D - W]\{T^4\} + \left[ \frac{1}{A} \right] [h - z]\{T\} + \left\{ \frac{q_{int}}{A} \right\} \end{aligned} \quad (22)$$

For the equilibrium case, the net heat flow to each node is zero, and with negligible conduction, equation (22) can be solved for  $\{T^4\}$ :

$$\sigma\{T^4\} = \left[ e(I - G_t e) + \frac{1}{\sigma}(D - W) \right]^{-1} \left\{ KS[a_s][I + G_s \rho_s]\{\mu_s\} + \rho_p S \cos^* \theta_s \sin^2 \omega [a_s][I + G_s \rho_s]\{\mu_p\} + \sigma T_p^4 \sin^2 \omega [e][I + G_t(I - e)]\{\mu_p\} + \left\{ \frac{q_{int}}{A} \right\} \right\} \quad (23)$$

For a multifaceted node which is nearly isothermal, although its faces receive different thermal radiation fluxes, the total net heat flow rate is given by

$$q_i = \sum_{j=1}^n \left( \frac{q}{A} \right)_j A_j$$

where  $n$  is the number of faces of node  $i$ . For the nonequilibrium case,

$$\frac{dT_i}{dt} = \frac{A_i}{m_i c_i} \left( \frac{q}{A} \right)_i \quad (24)$$

Equation (24) may be integrated numerically to yield the temperature history, beginning with given initial temperatures.

Listings of the three computer programs which calculate shape factors, projected areas, and temperatures are shown in appendix A. They are written in the FORTRAN IV language. The shape factors and projected areas are run in separate programs because they need to be calculated only once for each configuration. In the main program, surface properties, materials, heat loads, attitude, and flight trajectory can be varied for a fixed configuration without recalculating the shape factors and projected areas. Punched-card outputs from the other two programs supply shape factor and projected-area inputs to the main program.

#### Computation of Geometric Shape Factors and Projected Areas

Routines were written to enable the digital computer to calculate  $[F]$  and  $\{\mu\}$  using as input the coordinates of the corners of each plane quadrilateral into which the spacecraft has been divided. The factor which makes the calculation of the projected area of a flat plate complex is the shading by intercepting surfaces.

In order to calculate the unshaded projected area of a plane quadrilateral, it is divided into an  $n \times n$  grid. Formulas for the coordinates of the grid points and areas

of the grid elements are developed in appendix B. The ratio of the projected area to the total area is given by

$$\begin{aligned}\mu_i &= \frac{1}{A_i} \int_{A_i} (\bar{V}_{\text{source}} \cdot \bar{n}_i)^* I_{\text{sh},i} dA_i \\ &\approx \frac{1}{A_i} (\bar{V}_{\text{source}} \cdot \bar{n}_i) \sum_i I_{\text{sh},i} \Delta A_i\end{aligned}\quad (25)$$

where  $\bar{V}_{\text{source}}$  is the unit vector in the direction of the source and

$$I_{\text{sh}} = \begin{cases} 0 & \text{for shading} \\ 1 & \text{for no shading} \end{cases}$$

The method of determining shading is developed in appendix C.

In the computation of the shape factor between a pair of plane quadrilaterals, both are divided into  $n \times n$  grids as before, and each element of one is paired successively with each element of the other. The shading test is performed by the same scheme as for the projected areas. The shape factor  $F_{ij}$  between surfaces  $i$  and  $j$  is given by the average (taken over both surfaces) of the function  $\psi_{ij}$ :

$$\begin{aligned}F_{ij} &= \frac{1}{A_i A_j} \int_{A_i} \int_{A_j} \psi_{ij} dA_i dA_j \\ &\approx \frac{1}{A_i A_j} \sum_{i,j} \psi_{ij} \Delta A_i \Delta A_j \\ &\approx \frac{1}{A_i A_j} \sum_{i,j} \frac{-(\bar{n}_i \cdot \bar{r}_{ij})(\bar{n}_j \cdot \bar{r}_{ij})}{\pi r_{ij}^4} (I_{\text{sh}})_{ij} \Delta A_i \Delta A_j\end{aligned}\quad (26)$$

The position vector  $\bar{r}_{ij}$  is given by  $\bar{r}_j - \bar{r}_i$ , where  $\bar{r}_i$  and  $\bar{r}_j$  are the position vectors of the centroids of the elemental areas  $\Delta A_i$  and  $\Delta A_j$ . The same formulas as for the projected-area routine are used for the area of  $\Delta A_i$  and its centroid  $\bar{r}_i$ .

If a surface  $k$  is nonplanar and is approximated by a set of  $N$  planar surfaces, then  $\mu_k$  is

$$\mu_k = \frac{\sum_{i=1}^N \mu_i A_i}{\sum_{i=1}^N A_i} \quad (27)$$

and  $F_{kj}$  between the nonplanar surface and any other surface  $j$  is

$$F_{kj} = \frac{\sum_{i=1}^N A_i F_{ij}}{\sum_{i=1}^N A_i} \quad (28)$$

In this investigation the nodal surfaces into which a spacecraft is divided are called panels. A large member may be divided into several pieces to conform more nearly to the assumption of constant temperature over each panel. Surfaces of the spacecraft which can intercept thermal radiation which otherwise would impinge on any of the panels are called shaders. There will be fewer shaders than panels if any of the plane surfaces is subdivided into more than one panel.

#### EXAMPLE PROBLEM

An example of the application of the computer program is the prediction of temperatures on the proposed Meteoroid Technology Satellite. Figure 1 is a photograph of a model of the spacecraft shown attached to the last stage of its booster, with meteoroid-detector panels deployed. The cubical modules attached to the central structure are experiments for measuring velocities of meteoroids. The octagonal prism at the top is mainly solar cell area. In this example, the spacecraft is spinning about its axis of symmetry.

The projected areas and shape factors are calculated in separate programs to be used later as inputs to the temperature-prediction program. The Cartesian coordinates of panel corners and shader corners are used as input for the projected-area and shape factor programs. Actual areas of the panels are also provided by the projected-area program as input to the main program.

Direct inputs to the main program are absorptivities for solar radiation, hemispherical emissivities, the product of mass and specific heat for each panel, radiation

coefficients between back-to-back panels, and orbit and spacecraft-attitude parameters. A circular orbit with an altitude of 300 nautical miles (556 km) and an inclination of 38° to the equator was assumed.

In this example, 104 nodes were used. No conduction is accounted for in this example. Because of the small thicknesses and large surface areas of the nodes, radiation will outweigh conduction by far.

It was necessary to use a very small time increment (0.05 min) in the numerical integration of the temperatures. Some of the nodes consist of a sheet of plastic film 0.00025 inch (0.00064 cm) thick, with a very small thermal capacity; therefore, the solution is unstable for larger time increments.

The temperatures all converged within two or three orbits, without oscillation, to a solution which repeated itself on subsequent orbits. The temperatures compare reasonably with those obtained by a similar heat-transfer computer program which does not take multiple reflections into account.

The temperature history is plotted for seven nodes, whose locations are indicated in figure 1. The temperature plot for the outer face of one of the upper velocity detectors (see fig. 2) follows the heat inputs to it very closely because it has very low thermal inertia. At the far left of the plot, the decline from maximum value of the earth albedo and earth thermal fluxes is noted, while the solar input remains constant. The computed value of the albedo and earth thermal flux declines to zero at about 24 min after perigee, after which the albedo remains zero and the earth thermal flux increases. At about 31 min, the spacecraft enters the earth's shadow. During shadow, the earth thermal flux passes through a maximum and goes to zero again at about 71 min. Sunlight appears again at about 66 min. Albedo flux increases from zero at around 71 min to a maximum about midway through the sunlit period. The times of maximums and minimums in the thermal fluxes incident from external sources and in the temperature do not coincide exactly, since the node receives radiation from other parts of the spacecraft also.

Figures 3 and 4 show the temperature variation over the orbit for a horizontal detector and a vertical detector, respectively. A schematic illustration of the cross section of these meteoroid detectors is given in figure 5, with the bumper shields on each side shown.

#### CONCLUDING REMARKS

A method has been developed for the thermal analysis of geometrically complicated spacecraft whose surfaces can be approximated by plane quadrilaterals with gray surfaces having diffuse reflection and radiation properties. Optical blocking between surface

elements is accounted for automatically. Multiple diffuse reflections are also accounted for. Listings of the computer programs which perform the calculations are included.

Langley Research Center,  
National Aeronautics and Space Administration,  
Hampton, Va., May 1, 1970.

APPENDIX A  
COMPUTER PROGRAM LISTINGS  
Spacecraft Temperature Program

```

C PROGRAM ORBTEMP(INPUT,OUTPUT,TAPES=INPUT,TAPE6=OUTPUT) 0090
C HEAT TRANSFER PROGRAM FOR DIFFUSE RADIATION ON MULTIPANELED SPACECRAFT080
C
C DIMENSION AR1(104),AR2(104),AP1(104),APP1(104),APP2(104),CSH(104), 0070
C 1ETAS(104),ETAE(104),FSP1(104,104),FSPP1(104,104),FSPP2(104,104),
C 2HCOND(104,104),WATE(104),RRAD(104,104),T(104),XMU(104),TETAS(20),
C 3YMU(104),TXMU(104,20),TTXMU(20),DT(104),THELAM(3,3),URTH(3,3),
C 4           IPIVOT(104),INDEX(104,2)
C DIMENSION BTHETA(20),BALPHA(20),BBETA(20),RM3PN(3,3),RM1I(3,3),
C 1RM3OP(3,3),RM3A(3,3),RM1B(3,3),RM20(3,3),RM3NTH(3,3),RMOB(3,3),
C 2RMEB(3,3),UVEC(3,104),VEC(3)
C DIMENSION           XMUSUN(42, 7,19),TPHIS(36),PHIS(36),AXIS1(
C 13,50),AXIS2(3,50),TXMUS(360),VE1(3),VE2(3)
C DIMENSION VE3(3),RMIPHI(3,3),RMPIF(3,3)
C DIMENSION PHIE(104)
C 14 FORMAT(12F6.2) 0930
C 15 FORMAT(5E16.8) 0940
C 16 FORMAT(6F13.3) 0950
C 17 FORMAT(2X,10E12.5) 0960
C 18 FORMAT(//6H TIME=E15.8,18X,22HINTERNAL TEMPERATURE =E15.6/) 0970
C 19 FORMAT(1H1,9X,9HTIME(MIN),7X,14HTHETA(RADIANS),5X,15HALTITUDE(MILE
C 1S),5X,15HH 0=SHADE,1=SUN,9X,3HAMP,18X,7HETAS(1)//) 0980
C 23 FORMAT(1H1,10X,14HVALUES OF UVEC//) 0990
C 24 FORMAT(1H1,10X,14HPROGRAM INPUTS//) 1000
C 25 FORMAT(2X,6E20.6) 1010
C 1984 FORMAT(3(2I4,E16.8))
C 10101 FORMAT(7F11.8)
C 1856 FORMAT(7I10)
C 1905 FORMAT(8E16.8)
C
C NTIME=1
C READ(5,1856) NPANEL,NFAC,NHCOND,NRRAD,NINSFC,IFTLUP,ISPIN
C DO 3 I=1,NPANEL
C     AR2(I)=0.
C     APP2(I)=0.
C     DO 3 J=1,NPANEL
C         HCOND(I,J)=0.
C         FSP1(I,J)=0.
C         FSPP1(I,J)=0.
C 3     FSPP2(I,J)=0.
C
C     READ(5,1856) KEXT,KINT,KRRAD,KHCOND
C     READ(5,15)RADE,ALT,VELTO,VELRO,TI,DTI,TID,S
C     READ(5,15)TE,BOLT,G,AU,RE
C 1322 FORMAT(1H058HSOLAR CONSTANT,S-B CONSTANT,EARTH TEMP.,EARTH REFLECT 1030

```

## APPENDIX A -- Continued

```

11IVITY)
WRITE(6,1322)
WRITE(6,17) S,BOLT,TE,RE
READ(5,15)ABETA,AINCE,AINCO,ALPHA,COMEGA,DOMEGA,OMEGAP,PHIN,THETAO
IF(IFTLUP) 11,11,12
12 READ(5,15)BALPHA
READ(5,15)BBETA
READ(5,15)BTHETA
11 READ(5,15) (AR1(J),J=1,NPANEL)
READ(5,14) (AP1(J),J=1,NPANEL)
READ(5,14) (APP1(J),J=1,NPANEL)
READ(5,14) (CSH(J),J=1,NPANEL)
READ(5,14) (WATE(J),J=1,NPANEL)
WRITE(6,1313)
1313 FORMAT(1H010HEXT. AREAS)
WRITE(6,17) AR1
1314 FORMAT(1H025HEXT. SURF. ABSORPTIVITIES)
WRITE(6,1314)
WRITE(6,17) AP1
1316 FORMAT(1H023HEXT. SURF. EMISSIVITIES)
WRITE(6,1316)
WRITE(6,17) APP1
1318 FORMAT(1H07HWEIGHTS)
WRITE(6,1318)
WRITE(6,17) WATF
1320 FORMAT(1H014HSPECIFIC HEATS)
WRITE(6,1320)
WRITE(6,17) CSH
READ(5,14) (T(J),J=1,NPANEL)
IF(KEXT.EQ.0) GO TO 1111
DO 1492 K=1,NFAC
READ(5,1984) I1,J1,FSPP1(I1,J1),I2,J2,FSPP1(I2,J2),I3,J3,FSPP1(I3,
1J3)
FSPP1(J1,I1)=FSPP1(I1,J1)
FSPP1(J2,I2)=FSPP1(I2,J2)
1492 FSPP1(J3,I3)=FSPP1(I3,J3)
DO 1776 I=1,NPANEL
DO 1776 J=1,NPANEL
FSP1(I,J)=FSPP1(I,J)*AR1(J)
FSPP1(I,J)=FSP1(I,J)
RRAD(I,J)=-FSP1(I,J)*(1.-AP1(J))
1776 IF(I.EQ.J) RRAD(I,J)=RRAD(I,J)+1.
CALL SIMEQ(RRAD,NPANEL,FSP1,NPANEL,DETERM,IPIVOT,104,ISCALE)
1861 FORMAT(1H160X6HFSP1-S///)

```

## APPENDIX A – Continued

```

      WRITE(6,1861)
      WRITE(6,1905) ((FSPP1(J,K),K=1,NPANEL),J=1,NPANEL)
      DO 1849 I=1,NPANEL
      DO 1849 J=1,NPANEL
      RRAD(I,J)=-FSPP1(I,J)*(1.-APP1(J))
1849  IF(I.EQ.J) RRAD(I,J)=RRAD(I,J)+1.
      CALLSIMEQ(RRAD,NPANEL,FSPP1,NPANEL,DETERM,IPIVOT,104,ISCALE)
1862  FORMAT(1H160X7HFSPPP1-S///)
      WRITE(6,1862)
      WRITE(6,1905) ((FSPP1(J,K),K=1,NPANEL),J=1,NPANEL)
1111  CONTINUE
      RADO = RADE + ALT
      THETA = THETA0
      P = RADO*VELTO
      COSEGA=COS(COMEWA)
      SINEGA=SIN(COMEWA)
      COINCE=COS(AINCE)
      SINCE=SIN(AINCE)
      IF(KINT.EQ.0) GO TO 1033
      READ(5,14)(AR2(J),J=1,NPANEL)
      READ(5,14)(APP2(J),J=1,NPANEL)
      DO 7011 I=1,NINSFC
      READ(5,1984) I1,J1,FSPP2(I1,J1),I2,J2,FSPP2(I2,J2),I3,J3,FSPP2(I3,
1J3)
      FSPP2(J1,I1)=FSPP2(I1,J1)
      FSPP2(J2,I2)=FSPP2(I2,J2)
7011  FSPP2(J3,I3)=FSPP2(I3,J3)
      DO 1929 I=1,NPANEL
      DO 1929 J=1,NPANEL
      FSPP2(I,J)=FSPP2(I,J)*AR2(J)
      RRAD(I,J)=-FSPP2(I,J)*(1.-APP2(J))
1929  IF(I.EQ.J) RRAD(I,J)=RRAD(I,J)+1.
      CALLSIMEQ(RRAD,NPANEL,FSPP2,NPANEL,DETERM,IPIVOT,104,ISCALE)
1863  FORMAT(1H160X7HFSPPP2-S///)
      WRITE(6,1863)
      WRITE(6,1905) ((FSPP2(J,K),K=1,NPANEL),J=1,NPANEL)
1033  IF(KHCOND.EQ.0) GO TO 1030
      DO 1010 I=1,NHCOND
      READ(5,1984) I1,J1,HCOND(I1,J1),I2,J2,HCOND(I2,J2),I3,J3,HCOND(I3,J
13)
      HCOND(J1,I1)=HCOND(I1,J1)
      HCOND(J2,I2)=HCOND(I2,J2)
      HCOND(J3,I3)=HCOND(I3,J3)
1010  CONTINUE

```

## APPENDIX A – Continued

```

1324 FORMAT(1H123HCONDUCTION COEFFICIENTS)
      WRITE(6,1324)
      WRITE(6,17) HCOND
1030 CONTINUE
      DO 207 I=1,NPANEL
      DO 207 J=1,NPANEL
207  RRAD(I,J)=0.
      IF(KRRAD.EQ.0) GO TO 1020
      DO 1011 I=1,NRRAD
      READ(5,1984) I1,J1,RRAD(I1,J1),I2,J2,RRAD(I2,J2),I3,J3,RRAD(I3,J3)
      RRAD(J1,I1)=RRAD(I1,J1)
      RRAD(J2,I2)=RRAD(I2,J2)
      RRAD(J3,I3)=RRAD(I3,J3)
1011 CONTINUE
1020 CONTINUE
      READ(5,1856) NETAS,NPHIS
      IF(ISPIN) 1312,1312,1311
1311 DO 420 I=1,NPANEL
420  READ(5,10101) (TXMU(I,J),J=1,NETAS)
4891 FORMAT(1H012HMU BAR TABLE)
      WRITE(6,4891)
      WRITE(6,17) ((TXMU(I,J),J=1,NETAS),I=1,NPANEL)
1312 READ(5,14) (TETAS(M),M=1,NETAS)
4893 FORMAT(1H010HETAS TABLE)
      WRITE(6,4893)
      WRITE(6,17) (TETAS(M),M=1,NETAS)
      DO 7 M=1,NETAS
7   TETAS(M)=TETAS(M)*.0174532925
      IF(ISPIN.EQ.1) GO TO 103
532  DO 4 I=1,NPANEL
      DO 4 J=1,NETAS
4   READ(5,10101) (XMUSUN(I,J,K),K=1,NPHIS)
4895 FORMAT(1H120HMU(ETAS,PHIS) TABLES)
      WRITE(6,4895)
      DO 2849 I=1,NPANEL
      DO 2849 J=1,NETAS
2849 WRITE(6,1701) (XMUSUN(I,J,K),K=1,NPHIS)
1701 FORMAT(//2X,10E12.5)
      READ(5,14) (TPHIS(NP),NP=1,NPHIS)
4897 FORMAT(1H010HPHIS TABLE)
      WRITE(6,4897)
      WRITE(6,17) (TPHIS(NP),NP=1,NPHIS)
      DO 246 M=1,NPHIS
246  TPHIS(M)=TPHIS(M)*.01745329252

```

## APPENDIX A – Continued

```

531  READ(5,16) ((AXIS1(K,J),K=1,3),J=1,NPANEL)
      READ(5,16) ((AXIS2(K,J),K=1,3),J=1,NPANEL)
      READ(5,16) ((UVEC(K,J),K=1,3),J=1,NPANEL)
      DO 8 L=1,NPANEL
      STOR=AXIS1(1,L)
      AXIS1(1,L)=AXIS1(2,L)
      AXIS1(2,L)=AXIS1(3,L)
      AXIS1(3,L)=STOR
      STOR=AXIS2(1,L)
      AXIS2(1,L)=AXIS2(2,L)
      AXIS2(2,L)=AXIS2(3,L)
      AXIS2(3,L)=STOR
      STOR=UVEC(1,L)
      UVEC(1,L)=UVEC(2,L)
      UVEC(2,L)=UVEC(3,L)
      8  UVEC(3,L)=STOR
C      PROGRAM TO CALCULATE ENVIRONMENT OF SPACECRAFT      1610
C      1620
C
C      ROTMTX(M,N,ETA) COMPUTES THE ELEMENTS OF A ROTATION MATRIX M ABOUT AXIS N
C      (N=1 TO 3) THROUGH ANGLE ETA
C
103  CALL ROTMTX(RM3PN,3,PHIN)
      CALL ROTMTX(RM1I,1,AINCO)
      CALL ROTMTX(RM3OP,3,OMEGAP)
      CALL MULT(RMIPHI,RM1I,RM3PN)
      CALL MULT(RMPIF,RM3OP,RMIPHI)
100  IF(IFTLUP) 102,102,101
101  CALL FTLUP(THETA,ABETA,1,20,BTHETA,BBETA)      1660
      CALL FTLUP(THETA,ALPHA,1,20,BTHETA,BALPHA)      1670
102  SINTHE = RADE/RADO      1680
      COSTHE=(1.-SINTHE**2)**.5
      OMEGA=TI*DOMFGA
      THETAN=-THETA
      CALL ROTMTX(RM3NTH,3,THETAN)
      CALL ROTMTX(RM3A,3,ALPHA)
      CALL ROTMTX(RM1B,1,ABETA)
      CALL ROTMTX(RM20,2,OMEGA)
C
C      MULT(MP,M1,M2) MULTIPLIES 3X3 MATRICES M1 AND M2, GIVING (MP)=(M1)(M2)      1690
C
      CALL MULT(RMEB,RM3A,RMPIF)
      CALL MULT(RMEB,RM1B,RMEB)      2540
      CALL MULT(RMEB,RM20,RMEB)      2550

```

## APPENDIX A – Continued

```

C      RMEB IS COMPLETED          2560
C      ROTATION MATRIX RMEB ROTATES THE INERTIAL REFERENCE FRAME (Z-AXIS COINCIDENT
C      WITH PLANET SPIN AXIS, X-AXIS PASSES THROGH THE VERNAL EQUINOX) TO COINCIDE
C      WITH THE SPACECRAFT BODY AXES
C
C      DO 170 J=1,3
170  VEC(J)=RMEB(J,1)*COSEGA+RMEB(J,2)*SINEGA*COINCE+RMEB(J,3)*SINEGA*
     1 SINCE
     CALL MULT(RMOB,RM3A,RM3NTH)
     CALL MULT(RMOB,RM1B,RMOB)
     CALL MULT(RMOB,RM20,RMOB)
C
C      ROTATION MATRIX RMOB GIVES ORBIT POSITION. COLUMN 1 GIVES COMPONENTS OF UNIT
C      VECTOR IN DIRECTION OF CRAFT FROM CENTER OF PLANET
C
C      IF(ISPIN.EQ.1) GO TO 1
C      DO 160 I=1,NPANEL
C      DO 9 J=1,3
C      VE1(J)=AXIS1(J,I)
9     VE2(J)=AXIS2(J,I)
C      COSE=-UVEC(1,I)*RMOB(1,1)-UVEC(2,I)*RMOB(2,1)-UVEC(3,I)*RMOB(3,1)
C      SINE=(1.-COSE*COSE)**.5
C      IF(     COSE .LT..1E-7) ETAE(I)=1.57079633
C      IF(     COSE .GE..1E-7) ETAE(I)=ATAN2(SINE,COSE)
C      IF((1.-ABS(COSE)).LT.1.E-8) GO TO 1809
C      COSE=-RMOB(1,1)*VE1(1)-RMOB(2,1)*VE1(2)-RMOB(3,1)*VE1(3)
C      SINE=-RMOB(1,1)*VE2(1)-RMOB(2,1)*VE2(2)-RMOB(3,1)*VE2(3)
C      PHIE(I)=ATAN2(SINE,COSE)
C      GO TO 1810
1809 PHIE(I)=0.
1810 CONTINUE
C      COSE=UVEC(1,I)*VEC(1)+UVEC(2,I)*VEC(2)+UVEC(3,I)*VEC(3)
C      SINE=(1.-COSE*COSE)**.5
C      IF(     COSE .LT..1E-7) ETAS(I)=1.57079633
C      IF(     COSE .GE..1E-7) ETAS(I)=ATAN2(SINE,COSE)
C      IF((1.-ABS(COSE)).LT.1.E-8) GO TO 1918
C      COSE=VE1(1)*VEC(1)+VE1(2)*VEC(2)+VE1(3)*VEC(3)
C      SINE=VE2(1)*VEC(1)+VE2(2)*VEC(2)+VE2(3)*VEC(3)
C      PHIS(I)=ATAN2(SINE,COSE)
C      GO TO 160
1918 PHIS(I)=0.
160  CONTINUE
     GO TO 2
1     CONTINUE

```

2620

## APPENDIX A – Continued

```

COSE=VEC(2)
SINE=(1.-COSE*COSE)**.5
IF(ABS(COSE).LT..1E-7) ETASUN=1.57079633
IF(ABS(COSE).GE..1E-7) ETASUN=ATAN2(SINE,COSE)
COSE=-RMOB(2+1)
SINE=(1.-COSE*COSE)**.5
IF(ABS(COSE).LT..1E-7) ETARTH=1.57079633
IF(ABS(COSE).GE..1E-7) ETARTH=ATAN2(SINE,COSE)
DO 6 I=1,NPANEL
ETAS(I)=ETASUN
6 ETAE(I)=ETARTH
2 CONTINUE
AMP=RMOB(1+1)*VEC(1)+RMOB(2+1)*VEC(2)+RMOB(3+1)*VEC(3)
H=1.0
IF(-AMP.GT.COSTHE) H=0.0
IF(AMP.LT.0.0) AMP=0.0
2730
2740
2750
2920
C
DO 1000 J=1,NPANEL
IK=0
IF(ISPIN) 1357,1357,1358
1357 DO 680 I=1,NETAS
DO 680 K=1,NPHIS
IK=IK+1
680 TXMUS(IK)=XMUSUN(J,I,K)
NY=NETAS*NPHIS
CALL DISCOT(ETAS(J),PHIS(J),TETAS,TXMUS,TPHIS,22,NY,NPHIS,XMU(J))
CALL DISCOT(ETAF(J),PHIE(J),TETAS,TXMUS,TPHIS,22,NY,NPHIS,YMU(J))
GO TO 1000
1358 DO 1001 I=1,NETAS
1001 TTXMU(I)=TXMU(J,I)
CALL FTLUP(ETAS(J),XMU(J),2,NETAS,TETAS,TTXMU)
CALL FTLUP(ETAE(J),YMU(J),2,NETAS,TETAS,TTXMU)
1000 CONTINUE
* AT THIS POINT PROJECTED AREAS FOR SOLAR AND PARENT BODY ARE AVAILABLE
DO 1002 I=1,NPANEL
DT(I)=H*S*AR1(I)*XMU(I)*AP1(I)+BOLT*TE**4*SINTHE**2*AR1(I)*YMU(I)
1*APP1(I)+S*RE*AMP*SINTHE**2*AR1(I)*YMU(I)*AP1(I)
2750
C
C (DIR. SOLAR+DIR. PLANET EMISSION+ PLANET-REFLECTED SOLAR)
C
TEMP=0.
DO 1003 K=1,NPANEL
TEMP=TEMP+(APP1(K)*BOLT*T(K)**4+BOLT*TE**4*SINTHE**2*YMU(K)*(1.-
1*APP1(K))) *FSPP1(I,K)*AR1(I)*APP1(I)
2955

```

## APPENDIX A – Continued

```

1003 CONTINUE
C
C   (TEMP ADDS EXTERIOR LONG WAVELENGTH CONTRIB. TO PANEL I FROM ALL PANELS.
C       DIRECTLY AND BY ALL MULTIPLE DIFFUSE REFLECTIONS. CONSISTS OF
C           EXT. THERMAL EMISS. BY PANELS + REFLECTED PORTION OF PLANET EM
C
C   SFMP=0.
DO 1004 K=1,NPANEL
  SFMP=SFMP+(H*S*XMU(K)+S*RE*AMP*SINTHE**2*YMU(K))*(1.-AP1(K))           2975
  1 *FSP1(I,K)*AR1(I)*AP1(I)
1004 CONTINUE
C
C   (SEMP ADDS SOLAR SPECTRUM CONTRIB. FROM ALL PANELS INCL. ALL REFLECTIONS
C       CONSISTS OF PANEL-REFLECTED PORTION OF DIR. AND PLANET-REFL. SOLAR
C
C   UEMP=0.
DO 1005 K=1,NPANEL
  UEMP=UEMP+BOLT*APP2(K)*T(K)**4           *FSPP2(I,K)*AR2(I)*APP2(I)      2995
1005 CONTINUE
C
C   (UEMP ADDS INTERIOR LONG WAVELENGTH CONTRIB. TO I IF IT IS PART OF AN
C       ENCLOSURE . CONSISTS OF EMISSIONS AND ALL REFL. FROM SURF. OF
C           ENCLOSURE )
C
C   VEMP=0.
DO 1006 K=1,NPANEL
  VEMP=VEMP+HCOND(K,I)*(T(K)-T(I))
1006 CONTINUE
C
C   (VEMP IS THE CONDUCTION TO I )
C
C   WEMP=0.
DO 1007 K=1,NPANEL
  WEMP=WEMP+RRAD(K,I)*(T(K)**4-T(I)**4)
1007 CONTINUE
  WEMP=WEMP*AR1(I)
C
C   (WEMP GIVES RADIATION FROM ANOTHER PANEL PARALLEL TO I AND FORMING A FLAT
C       ENCLOSURE WITH I. EXAMPLE IS FLAT PRESSURE CELL METEOROID DETEC.
C
  XEMP=-BOLT*T(I)**4*(APP1(I)*AR1(I)+APP2(I)*AR2(I))
C
C   (XEMP GIVES LOSS BY EMISSION FROM I )
C

```

## APPENDIX A – Continued

```

DT(I)=DT(I)+TEMP+SEMP+UEMP+VEMP+WEMP+XEMP
DT(I)=DT(I)/WATE(I)/CSH(I)

1002 CONTINUE
DO 9843 I=1,NPANEL
9843 T(I)=T(I)+DT(I)*DTI
IF(MOD(NTIME,100).EQ.0)GO TO 113
GO TO 114
3100
113 CONTINUE
WRITE(6,17) (T(I),I=1,NPANEL)
114 IF(TI.LT.TID) GO TO 99
GO TO 999
3140
3150
C
C      ROUTINE TO CALCULATE RADIUS IN ELLIPTICAL ORBIT
C
99 CONTINUE
DDTI=.1*DTI
DO 992 J = 1,10
3160
3170
3180
3190
ACCEL = (P**2)/RADO**3 -G*(RADE/RADO)**2
ACCEL1 =ACCEL
DO 991 I = 1,5
3200
3220
3230
3250
RAD1=RADO+VELR0*DDTI+(2.*ACCEL+ACCEL1)*DDTI**2/6.
VELR1=VELR0+.5*(ACCEL+ACCEL1)*DDTI
991 ACCEL1 = (P**2)/RAD1**3 - G*(RADE/RAD1)**2
3290
RADH = RAD1-RADE
3300
ANVELO =P/RADO**2
3310
ANVEL1 = P/RAD1**2
3320
THETA = THETA + (ANVELO+ANVEL1)/2.0*(DTI/10.0)
3330
RADO = RAD1
3340
992 VELR0 = VELR1
3350
RATIO = RADH/ALT
3360
HIHT = RADH/5280.0
3380
TI=DTI*FLOAT(NTIME)
IF(MOD(NTIME,20).NE.0) GO TO 30
3390
WRITE(6,25) TI,THETA,HIHT,H,AMP,ETAS(1)
30 NTIME=NTIME+1
3400
GO TO 100
3410
999 STOP
3420
END
SUBROUTINE ROTMTX(ROTOR,M,ROTANG)
DIMENSION ROTOR(3,3)
ROTOR(M,M)=1.0
COSE=COS(ROTANG)
SINE=SIN(ROTANG)
DO 1 I=1,3

```

## APPENDIX A – Continued

```

IF(I.EQ.M) GO TO 1
ROTOR(I,M)=0.0
ROTOR(M,I)=0.0
ROTOR(I,I)=COS
1  CONTINUE
IF(M.EQ.1) GO TO 2
IF(M.EQ.2) GO TO 3
ROTOR(1,2)=SINE
ROTOR(2,1)=-SINE
RETURN
2  ROTOR(2,3)=SINE
ROTOR(3,2)=-SINE
RETURN
3  ROTOR(1,3)=-SINE
ROTOR(3,1)=SINE
RETURN
END
C                                         3430
C SUBROUTINE ROTMTX CALCULATES THE ELEMENTS OF A ROTATION MATRIX, GIVEN THE
C AXIS NO. AND THE CCW ANGLE OF ROTATION
SUBROUTINE MULT(A,B,C)                                         3450
DIMENSION A(3,3),B(3,3),C(3,3),D(3,3)
DO 1 I=1,3                                         3470
DO 1 J=1,3                                         3480
D(I,J)=0.0
DO 1 K=1,3
1  D(I,J)=D(I,J)+B(I,K)*C(K,J)
DO 2 I=1,3
DO 2 J=1,3
2  A(I,J)=D(I,J)
RETURN
END
                                         3550
                                         3560

```

## APPENDIX A - Continued

### Definitions and Instructions for the Temperature Program

```

C  HEAT TRANSFER PROGRAM FOR DIFFUSE RADIATION ON MULTIPANELED SPACECRAFT080
C 0090
C  SYMBOLS USED IN PROGRAM 0100
C
C     ABETA    ANGLE OUT OF ORBIT PLANE OF SPIN AXIS (YAW) 0140
C     AINCE    ANGLE BETWEEN ECLIPTIC PLANE AND EQUATORIAL PLANE 0150
C     AINCO    ORBIT INCLINATION ANGLE 0160
C     ALPHA    NEGATIVE ELEVATION ANGLE (PITCH) 0170
C     ALT      ALTITUDE OF SPACECRAFT (FT)
C
C     AMP      FOR DAYLIGHT, AMP=COS(ANGLE BETWEEN LINES TO THE SUN AND SPACECR
C               AFT DRAWN FROM THE CENTER OF THE EARTH) FOR DARKNESS,AMP=0
C
*     AP1(J)    ABSORPTIVITY OF EXTERNAL CAVITY TO P ENERGY, DIMENSIONLESS
*     APP1(J)    ABSORPTIVITY OF EXTERNAL CAVITY TO PP ENERGY, DIMENSIONLESS
*     APP2(J)    ABSORPTIVITY OF INTERNAL CAVITY TO PP ENERGY, DIMENSIONLESS
*     AR1(J)    EXTERNAL CAVITY AREA, INCH SQUARE
*     AR2(J)    INTERNAL CAVITY AREA, INCH SQUARE
C     AU       DISTANCE FROM SUN (ASTRONOMICAL UNITS) 0200
C     AXIS1    UNIT VECTOR IN PLANE OF PANEL
C     AXIS2    CROSS PRODUCT OF UVEC AND AXIS1
C
C     BALPHA   PITCH TABLE (20 PARTS) 0240
C     BBETA    YAW TABLE (20 PARTS) 0250
C     BTHETA   ORBIT ANGLE TABLE (20 PARTS) 0260
*     BOLT     STEFAN-BOLTZMANN CONSTANT
C     COMEGA   SUN ANGLE IN ELLIPTIC PLANE FROM VERNAL EQUINOX 0320
*     CSH(J)    SPECIFIC HEAT OF JTH COMPONENT, BTU/POUND/DEG
C     DDEL     TIME DERIVATIVE OF ORBIT ANGLE (RADIAN/SEC) 0340
C     DEL      ORBIT ANGLE (RADIAN) 0350
C     DOMEGA   SPIN RATE OF SPACECRAFT ABOUT 2-AXIS
C     DT(I)    TEMPERATURE INCREMENT OF THE ITH PANEL 0370
C     DTI      TIME INCREMENT (MIN) 0380
C     EPI      ANGLE OF SUN OUT OF ORBIT PLANE 0390
*     ETAE(J)   PARENT BODY ASPECT ANGLE
*     ETAS(J)   SOLAR ASPECT ANGLE
C
C     THE SHAPE FACTORS INPUT TO THE PROGRAM ARE ONLY GEOMETRIC. THE FACTORS
C     WHICH TAKE INTO ACCOUNT ALL REFLECTIONS AND ALL RE-REFLECTIONS ARE COMPUTED
C     IN THE PROGRAM USING GEOMETRIC FACTORS AND THE REFLECTIVITIES OF EACH NODE.
*     FSP1(K,J) EXTERNAL SHAPE FACTOR ELEMENT FOR P ENERGY, /INCH SQUARE
*     FSPP1(K,J) EXTERNAL SHAPE FACTOR ELEMENT FOR PP ENERGY, /INCH SQUARE
*     FSPP2(K,J) INTERNAL SHAPE FACTOR ELEMENT FOR PP ENERGY, /INCH SQUARE
C     G       ACCELERATION OF GRAVITY (FT/MIN**2) 0420
*     H       SUN INDICATOR, DIMENSIONLESS
*     HCOND(K,J) LINEAR HEAT CONDUCTION COEF, BTU/MIN/DEG
C     IFTLUP   INTEGER TO INDICATE USE OF BBETA, BTHETA, BALPHA
C               USE 0 FOR CONSTANTS AND 1 FOR TABLES 0490
C               USE 0 FOR CONSTANTS AND 1 FOR TABLES 0500
C     ISPIN IS ZERO FOR A STABILIZED VEHICLE, NON-ZERO FOR SPINNING.

```

## APPENDIX A – Continued

C KEXT,KINT,KRRAD, AND KCOND INDICATE WHETHER THERE ARE NON-ZERO ELEMENTS OF  
 C FSPP1/FSPP1,FSPP2,RRAD, AND HCOND RESPECTIVELY  
 C NETAS NO. OF VALUES IN TABLE OF ETAS  
 C NFAC IS THE NO. OF CARDS USED TO LOAD THE NON-ZERO GEOMETRIC SHAPE FACTORS  
 C BETWEEN EXTERNAL SURFACES(3 TO A CARD).FSPP1 IS INITIALLY USED AS A DUMMY  
 C NAME FOR THIS ARRAY.  
 C NHCOND IS SIMILARLY THE NO. OF CARDS FOR HCOND,NRRAD FOR RRAD,NINSFC FOR THE  
 C INTERNAL GEOM. SHAPE FACTORS(READ IN AS FSPP2).  
 C NPANEL NUMBER OF PANELS 0510  
 C NPHIS NO. OF VALUES IN TABLE OF PHIS  
 C OMEGA ROLL ANGLE OF BODY  
 C OMEGAP ARGUMENT OF PERIGEE 0530  
 C P DENOTES SOLAR SPECTRUM,PP DENOTES THERMAL RADIATION SPECTRUM PEAKING A  
 C T MUCH LONGER WAVELENGTHS  
 C PHIN ARGUMENT OF ASCENDING NODE 0560  
 C PHIS(J) SOLAR AZIMUTH ANGLE  
 C RADE EARTH RADIUS (FT) 0590  
 C RADH ALTITUDE OF SPACECRAFT (FT) 0600  
 C RADO INJECTION RADIUS OF ELLIPTICAL ORBIT (FT) 0610  
 \* REFE REFLECTIVITY OF PARENT BODY TO SOLAR ENERGY,DIMENSIONLESS  
 C RRAD(K,J) RADIATION COEFFICIENT,WITH NET HEAT TRANSFER RATE=  
 C AREA\*RRAD(K,J)\*(T(K)\*\*4-T(J)\*\*4)  
 \* S SOLAR CONSTANT,BTU/INCH SQUARE/MINUTE 0690  
 C SINTHE SINE OF THE EARTH VIEW ANGLE  
 \* T(J) TEMPERATURE OF JTH COMPONENT,DEG  
 \* TE TEMPERATURE OF PARENT BODY,DEG  
 C TETAS,TPHIS,TXMU,TXMUS TABLES OF ETAS,PHIS,XMU  
 C THETA ORBIT ANGLE FROM PERIGEE 0720  
 C THETAO INITIAL THETA (ARBITRARY) 0730  
 C TI TIME (MIN) 0760  
 C TID CALCULATION TIME LIMIT 0770  
 C UVEC UNIT NORMAL VECTOR TO PANEL  
 C VELRO INJECTION VELOCITY NORMAL TO THE EARTH (FT/MIN) 0810  
 C VELTO INJECTION VELOCITY TANGENTIAL TO THE EARTH(FT/MIN) 0820  
 \* WATE(J) MASS OF JTH COMPONENT,POUNDS  
 \* XMU(J) UNIT PROJECTED AREA,DIMENSIONLESS,SOLAR  
 C XMUSUN(K,I,J) TABLE OF XMU OF EACH PANEL FOR GIVEN ETAS AND PHIS  
 \* YMU(J) UNIT PROJECTED AREA,DIMENSIONLESS,PARENT BODY 0850  
 C  
 C ARRAY DIMENSIONS FOR AN INDIVIDUAL CASE ARE-  
 C BTHETA,BALPHA,BBETA-SIZE OF TABLES OF PITCH AND YAW VERSUS ORBIT ANGLE.  
 C UVEC(3,NPANEL)  
 C AR1,AR2,AP1,APP1,APP2,CSH,ETAS,ETAE-ALL (NPANEL).

## APPENDIX A – Continued

```
C FSP1,FSPP1,FSPP2,HCOND,RRAD,(NPANEL,NPANEL)•
C WATE,T,XMU,YMU-ALL (NPANEL)•
C TETAS(NETAS),TXMU(NPANEL,NETAS),TTXMU(NETAS),DT(NPANEL)•
C XMUSUN(NPANEL,NETAS,NPHIS),TPHIS(NPHIS),PHIS(NPHIS)•
C AXIS1(3,NPANEL),AXIS2(3,NPANEL),TXMUS(NETAS*NPHIS),FAR(NPANEL,NPANEL)•
C IPIVOT(NPANEL),INDEX(NPANEL+2)•
C
C
C INPUT DATA LOADING ORDER
C 1. (8I10) NPANEL,NFAC,NHCOND,NRRAD,NINSFC,IFTLUP,ISPIN•
C 2. (8I10) KEXT,KINT,KRRAD,KHCOND•
C 3. (5E16.8) RADE,ALT,VELTO,VELRO,TI / DTI,TID,S•
C 4. (5E16.8) TE,BOLT,G,AU,RE•
C 5. (5E16.8) ABETA,AINCE,AINCO,ALPHA,COMEGA/DOMEGA,OMEGAP,PHIN,THETAO•
C 6. IF(IFTLUP.GT.0),(5E16.8) BALPHA,BBETA,BTHETA•
C 7. (5E16.8) AR1 8. (12F6.2) AP1 9.(12F6.2)APP1 10.(12F6.2)CSH
C 11.(12F6.2)WATE 12.(12F6.2) T
C 13. IF(KEXT.GT.0) FSPP1 (3(2I4,E16.8))
C 14. IF(KINT.GT.0) (12F6.2) AR2/APP2 15. IF(KINT.GT.0) FSPP2 (3(2I4,E16.8)
C 16. IF(KHCOND.NE.0) HCOND (3(2I4,E16.8))
C 17. IF(KRRAD.NE.0) RRAD (3(2I4,E16.8))
C 18. (8I10) NETAS,NPHIS
C 19. (7F11.8) TXMU 20. (12F6.2) TETAS
C IF(ISPIN.EQ.0) - 21. (7F11.8) XMUSUN,22. (12F6.2) TPHIS, 23. (6F13.3)
C AXIS1,AXIS2,UVEC
C
C
C PROGRAM OUTPUT-
C 1. IF(KEXT.NE.0),(8E16.8) FSP1,FSPP1-CONVERTED TO ACCOUNT FOR REFLECTION
C 2. IF(KINT.NE.0),(8E16.8) FSPP2-ALSO CONVERTED
C 3. EACH MINUTE, TI(MIN.), THETA, ALT, (STAT.MI.), H(1. FOR SUNLIT), AMP(MEASURE OF PLANET-REFLECTED SUNLIGHT), ETAS(1). (2X,6E20.6)
C 4. EVERY TEN MINUTES, TEMP. OF EACH PANEL, DEG.R (2X,10E12.5)
```

## APPENDIX A – Continued

### Shape Factor Program

```

PROGRAM SHAPE (INPUT,OUTPUT,PUNCH,TAPE5=INPUT,TAPE6=OUTPUT)
C      SHAPE FACTOR PROGRAM FOR DIFFUSE RADIATION
C THE FOLLOWING INSTRUCTIONS MUST BE FOLLOWED WHEN USING THIS PROGRAM
C IN THE MAIN PROGRAM CHANGE DIMENSION STATEMENT TO READ
C      DIMENSION C(M,4,3),D(N,5,3),NUMB1(M),NUMB2(N),COEFFA(L,L,4),COEFFR(L,L,4)
C IN SUBROUTINE ZAP CHANGE DIMENSION STATEMENT TO READ
C      DIMENSION Z(N,5,3)
C IN SUBROUTINE AREA, CHANGE DIMENSIONS TO READ
C      DIMENSION COEFFA(L,L,4),COEFFR(L,L,4)
C IN SUBROUTINE BLOCK CHANGE DIMENSION STATEMENT TO READ
C      DIMENSION B(N,5,3),NUMB2(N)
C WHERE M IS THE NUMBER OF PANELS AND N IS THE NUMBER OF BLOCKING PANELS. L IS
C THE GRID SIZE.
C VARIABLE DEFINITIONS. NO IS THE NUMBER OF PANELS, NBLOCK IS THE NUMBER
C OF BLOCKING PANELS. C IS AN ARRAY CONTAINING THE COORDINATES OF PANELS.
C D IS AN ARRAY CONTAINING THE COORDINATES OF THE BLOCKING PANELS
C NUMB1 IS AN ARRAY OF THE NUMBERS ASSIGNED TO THE PLANES OF EACH PANEL
C NUMB2 IS AN ARRAY OF THE NUMBERS ASSIGNED TO THE PLANES OF EACH BLOCKER
C
C PUNCHED OUTPUT-
C      (3(2I4,E16.8)) NON-ZERO SHAPE FACTORS FOR PANELS I AND J,3 TO A CARD
C      I1,J1,SHPFAC(I1,J1),I2,J2,SHPFAC(I2,J2),I3,J3,SHPFAC(I3,J3)
C
C      DIMENSION C(100,4,3),D(75,5,3),A(4,3),B(4,3)
C      DIMENSION NUMB1(100),NUMB2(75)
C      DIMENSION TENT(3),NZ1(3),NZ2(3)
C      DIMENSION COEFFA(10,10,4),COEFFR(10,10,4)
C      COMMON/ONE/D
C      COMMON/TWO/NUMB2
C      COMMON/THREE/ COEFFA,COEFFR
86  FORMAT(7I10)
10  FORMAT(3E16.8,2I4)
11  FORMAT(12F6.1)
330 FORMAT(3(2I4,E16.8))
      READ(5,86) NO,NBLOCK
      READ(5,11)((C(I,J,K),K=1,3),J=1,4),I=1,NO)
      READ(5,11)((D(I,J,K),K=1,3),J=1,4),I=1,NBLOCK)
      READ(5,86)(NUMB1(I),I=1,NO)
      READ(5,86)(NUMB2(I),I=1,NBLOCK)
      WRITE(6,5)
5   FORMAT(1H 14HNO. OF PANELS,15HNO. OF BLOCKERS)
      WRITE(6,86) NO,NBLOCK
      WRITE(6,7)
7   FORMAT(1H017HPANEL COORDINATES)

```

## APPENDIX A – Continued

```

      WRITE(6,11) (((C(I,J,K),K=1,3),J=1,4),I=1,NO)
      WRITE(6,19)
19 FORMAT(1H 19HBLOCKER COORDINATES)
      WRITE(6,11) (((D(I,J,K),K=1,3),J=1,4),I=1,NBLOCK)
      WRITE(6,6)
6  FORMAT(1H016HPANEL PLANE NOS.)
      WRITE(6,86) (NUMB1(I),I=1,NO)
      WRITE(6,8)
8  FORMAT(1H018HBLOCKER PLANE NOS.)
      WRITE(6,86) (NUMB2(I),I=1,NBLOCK)
      WRITE(6,9)
9  FORMAT(1H14H    I4H    J3X9HARFA OF I7X9HAREA OF J)
NZ=1
NG=5
NG2=NG*NG
FOURN2=FLOAT(4*NG2)
TWON4=FLOAT(2*NG2*NG2)
DO 75 M=1,NG
I1=M-1
I21=I1+M
NI=NG-M
NI1=2*NI+1
DO 75 N=1,NG
J1=N-1
J21=J1+N
NJ=NG-N
NJ1=2*NJ+1
COEFFA(M,N,1)=FLOAT(NI*NJ+(NI+1)*(NJ+1))/TWON4
COEFFA(M,N,2)=FLOAT(I1*(NJ+1)+M*NJ)/TWON4
COEFFA(M,N,3)=FLOAT((NI+1)*J1+NI*N)/TWON4
COEFFA(M,N,4)=FLOAT(I1*J1+M*N)/TWON4
COEFFR(M,N,1)=FLOAT(NI1*NJ1)/FOURN2
COEFFR(M,N,2)=FLOAT(I21*NJ1)/FOURN2
COEFFR(M,N,3)=FLOAT(I21*J21)/FOURN2
75 COEFFR(M,N,4)=FLOAT(NI1*J21)/FOURN2
CALL ZAP(NBLOCK)
IX=NO-1
DO 3 I=1,IX
JX=I+1
DO 3 J=JX,NO
DO 1 K=1,3
DO 1 L=1,4
A(L,K)=C(I+L+K)
1 B(L,K)=C(J+L+K)

```

## APPENDIX A – Continued

```

NA=NUMB1(I)
NB=NUMB1(J)
IF(NA.EQ.NB) SAB=0.0
IF(NA.EQ.NB) GO TO 4
CALL FAKTOR(A,B,SAB,SDA,SDB,NA,NB,NG,NBLOCK,NG2)
4  CONTINUE
IF(SAB.GT.1.E-10)GO TO 333
IF((I.EQ.IX).AND.(J.EQ.NO)) GO TO 333
GO TO 3
333 TENT(NZ)=SAB
NZ1(NZ)=I
NZ2(NZ)=J
WRITE(6,213) I,J,SAB
213 FORMAT(2I4,E16.8,BH VIEWFAC)
NZ=NZ+1
IF(NZ.EQ.4) GO TO 335
IF((I.EQ.IX).AND.(J.EQ.NO)) GO TO 335
GO TO 3
335 PUNCH 330,NZ1(1),NZ2(1),TENT(1)+NZ1(2)+NZ2(2)+TENT(2)+NZ1(3)+NZ2(3)
1),TENT(3)
NZ=1
3  CONTINUE
STOP
END
SUBROUTINE ZAP(NBLOCK)
DIMENSION Z(75,5,3),C1(3),C2(3),C3(3),C4(3),C5(3),AM1(3),AM2(3),
1V(3)
COMMON/ONE/Z
I=1
5  DO 1 J=1,3
C2(J)=Z(I,2,J)-Z(I,1,J)
1  C3(J)=Z(I,4,J)-Z(I,1,J)
CALL CROSS(C5,C2,C3)
CALL CROSS(C1,C3,C5)
CALL DOT(DET,C1,C2)
CALL CROSS(AM2,C2,C5)
DO 2 K=1,3
AM1(K)=C1(K)/DET
2 AM2(K)=AM2(K)/DET
DO 3 J=1,3
3  V(J)=Z(I,1,J)+Z(I,3,J)-Z(I,2,J)-Z(I,4,J)
C4(1)=AM 1(1)*V(1)+AM 1(2)*V(2)+AM 1(3)*V(3)
C4(2)=AM 2(1)*V(1)+AM 2(2)*V(2)+AM 2(3)*V(3)
C4(3)=0.

```

## APPENDIX A – Continued

```

DO 6 J=1,3
Z(I,2,J)=C2(J)
Z(I,3,J)=C3(J)
Z(I,4,J)=C4(J)
6 Z(I,5,J)=C5(J)
I=I+1
IF(I.LT.(NBLOCK+1)) GO TO 5
RETURN
END
SUBROUTINE FAKTOR(A,B,SIJ,SDA,SDB,NA,NB,NG,NBLOCK,NG2)
DIMENSION A(4,3),B(4,3),DA(100),DB(100),RA(100,3),RB(100,3),AV(3),
1 BV(3),CV(3),DV(3),EV(3),FV(3),X(3),Y(3)
DIMENSION XA(3),CB(3),RIJ(16,3),RDOTN(16,2),GV(3),HV(3)
SIJ=0.0
I=1
DO 1 N=1,3
AV(N)=A(1,N)
BV(N)=A(2,N)
CV(N)=A(3,N)
DV(N)=B(1,N)
EV(N)=B(2,N)
FV(N)=B(3,N)
GV(N)=A(4,N)
HV(N)=B(4,N)
RIJ(1,N)=DV(N)-AV(N)
RIJ(2,N)=EV(N)-AV(N)
RIJ(3,N)=FV(N)-AV(N)
RIJ(4,N)=HV(N)-AV(N)
RIJ(5,N)=DV(N)-BV(N)
RIJ(6,N)=EV(N)-BV(N)
RIJ(7,N)=FV(N)-BV(N)
RIJ(8,N)=HV(N)-BV(N)
RIJ(9,N)=DV(N)-CV(N)
RIJ(10,N)=EV(N)-CV(N)
RIJ(11,N)=FV(N)-CV(N)
RIJ(12,N)=HV(N)-CV(N)
RIJ(13,N)=DV(N)-GV(N)
RIJ(14,N)=EV(N)-GV(N)
RIJ(15,N)=FV(N)-GV(N)
RIJ(16,N)=HV(N)-GV(N)
AV(N)=BV(N)-AV(N)
BV(N)=CV(N)-BV(N)
DV(N)=FV(N)-DV(N)
1 EV(N)=FV(N)-FV(N)

```

## APPENDIX A – Continued

```

CALL CROSS(CV,AV,BV)
CALL CROSS(FV,DV,EV)
DO 14 J=1,16
DO 14 N=1,2
14 RDOTN(J+N)=0.
DO 15 J=1,16
DO 13 N=1,3
  RDOTN(J,1)=RDOTN(J,1)+RIJ(J,N)*CV(N)
13 RDOTN(J,2)=RDOTN(J,2)+RIJ(J,N)*FV(N)
  IF((RDOTN(J,1).GE.0.1).AND.(RDOTN(J,2).LE.-0.1)) GO TO 5
15 CONTINUE
  SIJ=0.
  SDA=1.
  SDB=1.
  RETURN
5 CALL DOT(SA,CV,CV)
  CALL DOT(SB,FV,FV)
  SA=SA**.5
  SB=SB**.5
  DO 2 N=1,3
    CV(N)=CV(N)/SA
2  FV(N)=FV(N)/SB
  CALL AREA(A,DA,RA,NG,NG2)
  CALL AREA(B,DB,RB,NG,NG2)
  DO 3 L=1,NG2
  DO 3 J=1,NG2
  DO 4 N=1,3
    X(N)=RA(L,N)
    XA(N)=X(N)
    Y(N)=RB(J,N)
4  X(N)=X(N)-Y(N)
  CALL DOT(S1,CV,X)
  CALL DOT(S2,FV,X)
  IF((S1.GE.0.0).OR.(S2.LE.0.0)) GO TO 3
  CALL BLOKK(XA,Y,NBLOCK,G,NA,NB,I)
  IF(G.LT..5) GO TO 3
  CALL DOT(S3,X,X)
  S3=S3**2
  SIJ=-S1*S2/S3*DA(L)*DB(J)+SIJ
3  CONTINUE
11  SDA=0.0
  SDB=0.0
  DO 7 N=1,NG2
    SDA=SDA+DA(N)

```

APPENDIX A – Continued

```

7 SDB=SDB+DR(N)
SIJ=SIJ/(SDA*SDB*3.1415926536)
IF(((SIJ*SDA).LT.1.E-5).AND.((SIJ*SDB).LT.1.E-5)) SIJ=0.
RETURN
END
SUBROUTINE AREA(A,DA,RA,NG,NG2)
DIMENSION A(4,3),DA(100),RA(100,3),AV(3),BV(3),CV(3),DV(3)
1,T1(3),T2(3),T3(3),T4(3),T5(3),T6(3),T7(3),T8(3)
DIMENSION COEFFA(10,10,4),COEFFR(10,10,4)
COMMON /THREE/ COEFFA,COEFFR
18 FORMAT(4E16.8)
DO 2 N=1,3
  AV(N)=A(1,N)
  BV(N)=A(2,N)
  CV(N)=A(3,N)
  DV(N)=A(4,N)
  T1(N)=BV(N)-AV(N)
  T2(N)=CV(N)-BV(N)
  T3(N)=CV(N)-DV(N)
2  T4(N)=DV(N)-AV(N)
  CALL CROSS(T5,T4,T1)
  CALL CROSS(T6,T2,T1)
  CALL CROSS(T7,T4,T3)
  CALL CROSS(T8,T2,T3)
  CALL DOT(Z1,T5,T5)
  CALL DOT(Z2,T6,T6)
  CALL DOT(Z3,T7,T7)
  CALL DOT(Z4,T8,T8)
  Z1=Z1**.5
  Z2=Z2**.5
  Z3=Z3**.5
  Z4=Z4**.5
  K=1
  DO 1967 I=1,NG
  DO 1967 J=1,NG
    DA(K)=Z1*COEFFA(I,J,1)+Z2*COEFFA(I,J,2)+Z3*COEFFA(I,J,3)+Z4*COEFFA
1(I,J,4)
    DO 3 M=1,3
3  RA(K,M)=AV(M)*COEFFR(I,J,1)+BV(M)*COEFFR(I,J,2)+CV(M)*COEFFR(I,J,3)
   1+DV(M)*COEFFR(I,J,4)
1967 K=K+1
  RETURN
END
SUBROUTINE BLOKK(UV,VV,NBLOCK,G,NU,NV,I)

```

## APPENDIX A – Continued

```

DIMENSION UV(3),VV(3),B(75,5,3),CI2(3),CI3(3),CI5(3)
DIMENSION NUMB2(75),AM(3),AMI1(3),AMI2(3),AMI3(3),RIMINA(3)
COMMON/ONE/B
COMMON/TWO/NUMB2
EPI=1.0E-06
IJ=1
CALL SUB(AM,UV,VV)
3 CONTINUE
IF((NU.EQ.NUMB2(I)).OR.(NV.EQ.NUMB2(I)))GO TO 100
DO 7 L=1,3
CI2(L)=B(I,2,L)
CI5(L)=B(I,5,L)
7 CI3(L)=B(I,3,L)
CALL DOT(DET,AM,CI5)
IF(DET**2.LT.EPI) GO TO 100
CALL CROSS(AMI1,AM,CI2)
CALL CROSS(AMI2,CI3,AM)
DO 8 M=1,3
AMI1(M)=AMI1(M)/DET
AMI2(M)=AMI2(M)/DET
AMI3(M)=B(I,5,M)/DET
8 RIMINA(M)=UV(M)-B(I,1,M)
CALL DOT(V3,AMI3,RIMINA)
IF((V 3 .GE.1.0).OR.(V 3 .LE.0.)) GO TO 100
CALL DOT(V1,AMI1,RIMINA)
IF(V 1 .LT.0.0) GO TO 100
CALL DOT(V2,AMI2,RIMINA)
IF(V 2 .LT.0.0) GO TO 100
IF(((V 2 -1.0)*(1.0+B(I,4,2))-V 1 *B(I,4,1)).GT.0.0) GO TO 100
IF(((V 1 -1.0)*(1.0+B(I,4,1))-V 2 *B(I,4,2)).GT.0.0) GO TO 100
G=0.0
GO TO 60
100 CONTINUE
IF(IJ.EQ.NBLOCK) GO TO 98
IF(I.EQ.NBLOCK) GO TO 10
I=I+1
GO TO 11
10 I=1
11 IJ=IJ+1
GO TO 3
98 G=1.0
60 CONTINUE
RETURN
END

```

## APPENDIX A – Continued

```
SUBROUTINE SUB(C,A,B)
DIMENSION C(3),A(3),B(3)
DO 1 J=1,3
1 C(J)=A(J)-B(J)
RETURN
END
SUBROUTINE CROSS(C,A,B)
DIMENSION C(3),A(3),B(3)
C(1)=A(2)*B(3)-B(2)*A(3)
C(2)=B(1)*A(3)-A(1)*B(3)
C(3)=A(1)*B(2)-B(1)*A(2)
RETURN
END
SUBROUTINE DOT(C,A,B)
DIMENSION A(3),B(3)
C=A(1)*B(1)+A(2)*B(2)+A(3)*B(3)
RETURN
END
```

## APPENDIX A – Continued

### Projected Area Program

```

PROGRAM PROJAR(INPUT,OUTPUT,TAPE5=INPUT,TAPE6=OUTPUT,PUNCH)
C   PROGRAM FOR PROJECTED AREAS OF PLANE QUADRILATERALS, ACCOUNTING FOR SHADING
C   ARRAY DIMENSIONS WILL BE AS FOLLOWS-
C MAIN PROGRAM- A(NPNL,4,3),B(NSHDR,5,3),NUMP(NPNL),NUMS(NSHDR),ETAS(37),
C   PHIS(NN),RETAS(37),RPHIS(72),ARPAN(NPNL),XMUAV(NPNL,3),AND UNRMVEC(NPNL,3).
C SUBROUTINE PLANE-Z(NSHDR,5,3)
C SUBROUTINE SHADE-B(NSHDR,5,3),NUMB2(NSHDR)
C   WHERE NPNL IS THE NUMBER OF PANELS,NSHDR IS THE NUMBER OF SHADERS,NM IS
C THE NUMBER OF VALUES OF ETA-S,AND NN IS THE NUMBER OF VALUES OF PHI-S.
C   IN THE PROJECTED AREA PROGRAM,NG IS THE NUMBER OF EQUAL SEGMENTS INTO
C WHICH EACH SIDE OF EACH PANEL IS DIVIDED TO FORM A GRID OF ELEMENTAL AREAS.
C IF NG IS DIFFERENT FROM 10,THE DIMENSIONS MUST BE DA(NG**2),RG(NG**2,3),
C COEFFA(NG,NG,4),COEFFR(NG,NG,4) IN THE PROGRAM-AND IN THE SUBROUTINE AREA.
C DA(NG**2),RA(NG**2,3),COEFFA(NG,NG,4),AND COEFFR(NG,NG,4).
C   DEFINITIONS OF VARIABLES-
C   COEFFA(M,N,K) IS ONE OF THE FOUR COEFFICIENTS IN THE FORMULA FOR THE AREA
C   OF GRID ELEMENT M,N OF A PANEL
C   COEFFR(M,N,K) IS THE COEFFICIENT OF THE COORDINATES OF THE KTH CORNER OF THE
C   PANEL IN THE FORMULA FOR THE CENTROID OF ELEMENT M,N
C NUMP IS AN ARRAY OF NUMBERS IDENTIFYING THE PLANE OF EACH PANEL
C NUMS IS A SIMILAR ARRAY FOR THE SHADERS
C A(I,J,K)GIVES THE CARTESIAN COORDINATES OF CORNER J(J=1 TO 4) OF PANEL I
C K=1 TO 3 CORRESPONDS TO X,Y,AND Z COORDINATES RESPECTIVELY.
C B GIVES THE SHADER COORDINATES.
C ETAS IS THE ARRAY OF ETA-S VALUES,PHIS OF THE PHI-S VALUES.
      DIMENSION A(50,4,3),B(60,5,3),NUMP(50),NUMS(60),ETAS(37)
      1,PHIS(72),C(4,3),C1(3),C2(3),C3(3),C4(3),VNORM(3),V2(3),SUN(3),
      2,RETAS(37),RPHIS(72),DA(100),RG(100,3),RSUN(3),RGR(3)
      DIMENSION COEFFA(10,10,4),COEFFR(10,10,4)
      DIMENSION ARPAN(50),XMUAV( 19 ),UNRMVEC(50,3)
      DIMENSION VEE1(3),VEF2(3)
      DIMENSION XMUPHI(72)
      DIMENSION AXIS1(50,3),AXIS2(50,3)
      DIMENSION CSET(37),SNET(37),SNPH(72),CSPH(72)
      COMMON /ONE/B /TWO/NUMS
      COMMON /THREF/ COEFFA,COEFFR
30  FORMAT(7I10)
31  FORMAT(12F6.1)
      READ(5,30) NPNL,NSHDR,ISPIN
      READ(5,30) NG
      READ(5,30) (NUMP(I),I=1,NPNL)
      READ(5,30) (NUMS(I),I=1,NSHDR)
      READ(5,31) (((A(I,J,K),K=1,3),J=1,4),I=1,NPNL)
      READ(5,31) (((B(I,J,K),K=1,3),J=1,4),I=1,NSHDR)

```

## APPENDIX A – Continued

```

      WRITE(6,32)
32  FORMAT(1H 10HNO. PANELS11HNO. SHADERS)
      WRITE(6,30) NPNL,NSHDR
      WRITE(6,34)
34  FORMAT(1H012HSIZE OF GRID)
      WRITE(6,30) NG
      WRITE(6,35)
35  FORMAT(1H016HPANEL PLANE NOS.)
      WRITE(6,30) (NUMP(I),I=1,NPNL)
      WRITE(6,36)
36  FORMAT(1H017HSHADER PLANE NOS.)
      WRITE(6,30) (NUMS(I),I=1,NSHDR)
      WRITE(6,37)
37  FORMAT(1H017HPANEL COORDINATES)
97  FORMAT(12F7.1)
      WRITE(6,97) (((A(I,J,K),K=1,3),J=1,4),I=1,NPNL)
      WRITE(6,38)
38  FORMAT(1H018HSHADER COORDINATES)
      WRITE(6,97) (((B(I,J,K),K=1,3),J=1,4),I=1,NSHDR)
      NG2=NG*NG
      FOURN2=FLOAT(4*NG2)
      TWON4=FLOAT(2*NG2*NG2)
      DO 75 M=1,NG
      I1=M-1
      I21=I1+M
      NI=NG-M
      NI1=2*NI+1
      DO 75 N=1,NG
      J1=N-1
      J21=J1+N
      NJ=NG-N
      NJ1=2*NJ+1
      COEFFA(M,N,1)=FLOAT(NI*NJ+(NI+1)*(NJ+1))/TWON4
      COEFFA(M,N,2)=FLOAT(I1*(NJ+1)+M*NJ)/TWON4
      COEFFA(M,N,3)=FLOAT((NI+1)*J1+NI*N)/TWON4
      COEFFA(M,N,4)=FLOAT(I1*NJ1+M*N)/TWON4
      COEFFR(M,N,1)=FLOAT(NI1*NJ1)/FOURN2
      COEFFR(M,N,2)=FLOAT(I21*NJ1)/FOURN2
      COEFFR(M,N,3)=FLOAT(I21*NJ1)/FOURN2
75  COEFFR(M,N,4)=FLOAT(NI1*NJ1)/FOURN2
      CALL PLANE(NSHDR)
51  READ(5,30) NM,NN
      IF (EOF,5)998+999
998 STOP

```

## APPENDIX A – Continued

```

999 CONTINUE
  READ(5,31) (ETAS(I),I=1,NM)
  READ(5,31) (PHIS(I),I=1,NN)
  WRITE(6,33)
33  FORMAT(1H012HNO. OF ETA-S12HNO. OF PHI-S)
  WRITE(6,30) NM,NN
  WRITE(6,39)
39  FORMAT(1H014HETA-SUN VALUES)
  WRITE(6,31) (ETAS(I),I=1,NM)
  WRITE(6,40)
40  FORMAT(1H014HPHI-SUN VALUES)
  WRITE(6,31) (PHIS(I),I=1,NN)
  TOT=FLOAT(NN)
  ISHDR=1
  DO 150 LM=1,NM
  RETAS(LM)=ETAS(LM)*1.745329252E-2
  CSET(LM)=COS(RETAS(LM))
150  SNET(LM)=SIN(RETAS(LM))
  DO 175 MN=1,NN
  RPHIS(MN)=PHIS(MN)*1.745329252E-2
  SNPH(MN)=SIN(RPHIS(MN))
175  CSPH(MN)=COS(RPHIS(MN))
  DO 1 IP=1,NPNL
  NI=NUMP(IP)
  DO 2 JA=1,3
  DO 8 JC=1,4
8   C(JC,JA)=A(IP,JC,JA)
  C1(JA)=C(2,JA)-C(1,JA)
  C2(JA)=C(4,JA)-C(1,JA)
  C3(JA)=C(4,JA)-C(3,JA)
2   C4(JA)=C(2,JA)-C(3,JA)
  CALL CROSS(VNORM,C1,C2)
  CALL CROSS(V2,C3,C4)
  CALL DOT(D,VNORM,VNORM)
  CALL DOT(D1,V2,V2)
  D=D**.5
  DO 68 JQ=1,3
68  UNRMVEC(IP+JQ)=VNORM(JQ)/D
  D1=D1**.5
  ARMAC=.5*(D+D1)
  ARPAN(IP)=ARMAC
  CALL AREA(C,DA,RG,NG,NG2)
  ARTOT=0.
  DO 13 J=1,NG2

```

APPENDIX A – Continued

```

13 ARTOT=ARTOT+DA(J)
  WRITE(6,16) IP
16 FORMAT(1H111H PANEL NO.=I2)
  WRITE(6,15) ARMAC
15 FORMAT(1H020X,5HAREA=E16.8,2X,7HSQ. IN.)
  IF((ABS(ARTOT/ARMAC-1.0).GE..01) WRITE(6,14)
14 FORMAT(1H037HAREAS DO NOT AGREE WITHIN ONE PERCENT)
  WRITE(6,25) ARTOT
25 FORMAT(1H 23HSUM OF ELEMENTAL AREAS=E16.8,2X,7HSQ. IN.)
  DO 3 KN=1,3
 3 VNORM(KN)=VNORM(KN)/D
  DO 4 LM=1,NM
  SUM=0.
  WRITE(6,17)
17 FORMAT(1H03X4HETAS4X4HPHIS3X6HARPROJ10X2HMU)
  DO 5 MN=1,NN
  ARPROJ=0.
  IF(ISPIN.EQ.1) GO TO 27
  GO TO 21
27 SUN(1)=SNET(LM)*CSPH(MN)
  SUN(2)=SNET(LM)*SNPH(MN)
  SUN(3)=CSET(LM)
  CALL DOT(SDOTN,VNORM,SUN)
  IF(SDOTN.LE.0.) GO TO 23
  GO TO 47
21 CONTINUE
  DO 29 K=1,3
29 VEE1(K)=C1(K)
  SDOTN=CSET(LM)
  CALL DOT(VV1+VEE1+VEF1)
  VV1=VV1**.5
  DO 24 K=1,3
24 VEE1(K)=VEE1(K)/VV1
  CALL CROSS(VEE2,VNORM,VEE1)
  DO 26 K=1,3
  AXIS1(IP,K)=VEE1(K)
  AXIS2(IP,K)=VEE2(K)
26 SUN(K)=VNORM(K)*CSET(LM)+VEE1(K)*SNET(LM)*CSPH(MN)+VEE2(K)*SNET(LM)
  1)*SNPH(MN)
47 DO 7 N=1,3
 7 SUN(N)=1000.*SUN(N)
  DO 9 J=1,NG2
  DO 10 K=1,3
  RSUN(K)=RG(J,K)+SUN(K)

```

## APPENDIX A – Continued

```

10 RGR(K)=RG(J,K)
   CALL SHADE(RGR,RSUN,NSHDR,H,NI,ISHDR)
   IF(H.LT..5) GO TO 9
   ARPROJ=ARPROJ+DA(J)
9  CONTINUE
   ARPROJ=ARPROJ*SDOTN
23  XMU=ARPROJ/ARTOT
   WRITE(6,18) ETAS(LM),PHIS(MN),ARPROJ+XMU
   XMUPHI(MN)=XMU
18  FORMAT(2F8.1,2E16.8)
   IF(ISPIN) 5,5,100
100 SUM=SUM+ARPROJ
5   CONTINUE
   IF(ISPIN) 77,77,78
77  PUNCH 88,(XMUPHI(LP),LP=1,NN)
   GO TO 4
78  ARPJBR=SUM/TOT
   XMUBAR=ARPJBR/ARTOT
   XMUAV(LM)=XMUBAR
   WRITE(6,19)
19  FORMAT(1H015HMFAN PROJ. AREA,7X,10HAVERAGE MU)
   WRITE(6,20) ARPJBR,XMUBAR
20  FORMAT(2E16.8)
4   CONTINUE
   IF(ISPIN.EQ.0) GO TO 1
   PUNCH 88,(XMUAV(LM),LM=1,NM)
1   CONTINUE
88  FORMAT(7F11.8)
49  FORMAT(5E16.9)
   PUNCH 49,(ARPAN(IP),IP=1,NPNL)
50  FORMAT(6F13.9)
   IF(ISPIN) 72,72,73
72  PUNCH 50,((UNRMVEC(IP,JQ),JQ=1,3),IP=1,NPNL)
   PUNCH 50,((AXIS1(IP+K),K=1,3),IP=1,NPNL)
   PUNCH 50,((AXIS2(IP+K),K=1,3),IP=1,NPNL)
73  PUNCH 30, NM,NN
   PUNCH 31,(ETAS(M),M=1,NM)
   PUNCH 31,(PHIS(N),N=1,NN)
   GO TO 51
END
SUBROUTINE PLANE(NBLOCK)
DIMENSION Z(60,5,3),C1(3),C2(3),C3(3),C4(3),C5(3),AM1(3),AM2(3),
1V(3)
COMMON/ONE/Z

```

## APPENDIX A – Continued

```

I=1
5  DO 1 J=1,3
C2(J)=Z(I,2,J)-Z(I,1,J)
1  C3(J)=Z(I,4,J)-Z(I,1,J)
CALL CROSS(C5,C2,C3)
CALL CROSS(C1,C3,C5)
CALL CROSS(AM2,C5,C2)
CALL DOT(DET,C1,C2)
DO 2 K=1,3
AM1(K)=C1(K)/DET
2 AM2(K)=AM2(K)/DET
DO 3 J=1,3
3  V(J)=Z(I,1,J)+Z(I,3,J)-Z(I,2,J)-Z(I,4,J)
C4(1)=AM 1(1)*V(1)+AM 1(2)*V(2)+AM 1(3)*V(3)
C4(2)=AM 2(1)*V(1)+AM 2(2)*V(2)+AM 2(3)*V(3)
C4(3)=0.
DO 6 J=1,3
Z(I,2,J)=C2(J)
Z(I,3,J)=C3(J)
Z(I,4,J)=C4(J)
6  Z(I,5,J)=C5(J)
I=I+1
IF(I.LT.(NBLOCK+1)) GO TO 5
RETURN
END
C  SUBR. PLANE OPERATES ON ARRAY Z (BODY COORDINATES OF THE 4 CORNERS OF EACH
C  SHADER) TO FIND PARAMETERS NEEDED BY THE SHADING SUBR. (SHADE)
SUBROUTINE AREA(A,DA,RA,NG,NG2)
DIMENSION A(4,3),DA(100),RA(100,3),AV(3),BV(3),CV(3),DV(3)
1.T1(3),T2(3),T3(3),T4(3),T5(3),T6(3),T7(3),T8(3)
DIMENSION COEFFA(10,10,4),COEFFR(10,10,4)
COMMON /THREE/ COEFFA,COEFFR
18  FORMAT(4E16.8)
DO 2 N=1,3
AV(N)=A(1,N)
BV(N)=A(2,N)
CV(N)=A(3,N)
DV(N)=A(4,N)
T1(N)=BV(N)-AV(N)
T2(N)=CV(N)-BV(N)
T3(N)=CV(N)-DV(N)
2 T4(N)=DV(N)-AV(N)
CALL CROSS(T5,T4,T1)
CALL CROSS(T6,T2,T1)

```

## APPENDIX A – Continued

```

CALL CROSS(T7,T4,T3)
CALL CROSS(T8,T2,T3)
CALL DOT(Z1,T5,T5)
CALL DOT(Z2,T6,T6)
CALL DOT(Z3,T7,T7)
CALL DOT(Z4,T8,T8)
Z1=Z1**.5
Z2=Z2**.5
Z3=Z3**.5
Z4=Z4**.5
K=1
DO 1967 I=1,NG
DO 1967 J=1,NG
  DA(K)=Z1*COEFFA(I,J,1)+Z2*COEFFA(I,J,2)+Z3*COEFFA(I,J,3)+Z4*COEFFA
  1(I,J,4)
  DO 3 M=1,3
3  RA(K,M)=AV(M)*COEFFR(I,J,1)+BV(M)*COEFFR(I,J,2)+CV(M)*COEFFR(I,J,3
  1)+DV(M)*COEFFR(I,J,4)
1967 K=K+1
      RETURN
      END
C  SUBR. AREA,USING THE COORDINATES OF EACH CORNER AND THE ARRAYS COEFFA AND
C  COEFFR,COMPUTES THE COORDINATES OF THE CENTROID AND THE AREA OF EACH GRID
C  ELEMENT OF A PANEL.
SUBROUTINE SHADE(UV,VV,NBLOCK,G,NU,1)
DIMENSION UV(3),VV(3),B(60,5,3),CI2(3),CI3(3),CI5(3)
DIMENSION NUMB2(60),AM(3),AMI1(3),AMI2(3),AMI3(3),AMINX(3)
COMMON/ONE/B
COMMON/TWO/NUMB2
EPI=1.0E-06
IJ=1
CALL SUB(AM,VV,UV)
3  CONTINUE
  IF (NU.EQ.NUMB2(1))GO TO 100
  DO 7 L=1,3
    CI2(L)=B(I,2,L)
    CI5(L)=B(I,5,L)
7  CI3(L)=B(I,3,L)
    CALL DOT(DET,AM,CI5)
    IF(DET**2.LT.EPI) GO TO 100
    CALL CROSS(AMI1,AM,CI3)
    CALL CROSS(AMI2,CI2,AM)
    DO 8 M=1,3
      AMI1(M)=AMI1(M)/DET

```

## APPENDIX A – Concluded

```

AMI2(M)=AMI2(M)/DET
AMI3(M)=B(I,5,M)/DET
8  AMINX(M)=B(I+1,M)-UV(M)
CALL DOT(V3,AMI3,AMINX)
IF((V 3 .GE.1.0).OR.(V 3 .LE.0.)) GO TO 100
CALL DOT(V1,AMI1,AMINX)
IF(V 1 .LT.0.0) GO TO 100
CALL DOT(V2,AMI2,AMINX)
IF(V 2 .LT.0.0) GO TO 100
IF(((V1-1.)*(1.+B(I+4,2))-V2*B(I+4,1)).GT.0.) GO TO 100
IF(((V2-1.)*(1.+B(I+4,1))-V1*B(I+4,2)).GT.0.) GO TO 100
G=0.0
GO TO 60
100 CONTINUE
IF(IJ.EQ.NBLOCK) GO TO 98
IF(I.EQ.NBLOCK) GO TO 10
I=I+1
GO TO 11
10 I=1
11 IJ=IJ+1
GO TO 3
98 G=1.0
60 CONTINUE
RETURN
END
C  SUBR. SHADE DETERMINES WHETHER THE LINE BETWEEN TWO GIVEN POINTS INTERSECTS
C  ANY OF THE SHADERS.
SUBROUTINE SUB(C,A,B)
DIMENSION A(3),B(3),C(3)
DO 1 I=1,3
1  C(I)=A(I)-B(I)
RETURN
END
C  SUBR. SUB GIVES VECTOR C=B-A
SUBROUTINE CROSS(C,A,B)
DIMENSION A(3),B(3),C(3)
C(1)=A(2)*B(3)-A(3)*B(2)
C(2)=A(3)*B(1)-A(1)*B(3)
C(3)=A(1)*B(2)-A(2)*B(1)
RETURN
END
C  SUBR. CROSS GIVES VECTOR C=CROSS PRODUCT OF VECTORS A AND B
SUBROUTINE DOT(C,A,B)
DIMENSION A(3),B(3)

C=A(1)*B(1)+A(2)*B(2)+A(3)*B(3)
RETURN
END
C  SUBR. DOT GIVES C=DOT PRODUCT OF VECTORS A AND B.

```

APPENDIX B

DERIVATION OF CENTROID AND AREA FORMULAS  
FOR ELEMENTAL GRID SECTIONS OF A  
PLANE QUADRILATERAL

The plane quadrilateral is divided into an  $n \times n$  grid by dividing each side into  $n$  equal segments and connecting the corresponding dividing points on opposite sides. (See fig. 6.) Counterclockwise from the lower left, the position vectors of the vertexes are  $\bar{A}$ ,  $\bar{B}$ ,  $\bar{C}$ , and  $\bar{D}$ . Any vertex may be taken as the starting point, so long as the order is counterclockwise. This gives the proper sense to the computed normal vector to the plane.

The lower left corner of each elemental area is denoted by  $(j, k)$ , where  $j$  and  $k$  are integers increasing from 1 to  $n$ ;  $j$ , from left to right; and  $k$ , from bottom to top. Now, each segment of the line between  $\bar{A}$  and  $\bar{B}$  represents a change in position of  $\frac{\bar{B} - \bar{A}}{n}$ . Thus, the position vector of the point  $(j, 1)$  is given by

$$\bar{V}_{j,1} = \bar{A} + (j - 1) \frac{\bar{B} - \bar{A}}{n} = \frac{1}{n} [(n - j + 1)\bar{A} + (j - 1)\bar{B}] \quad (B1)$$

Similarly, along the line from  $\bar{D}$  to  $\bar{C}$  (where  $k = n + 1$ ),

$$\bar{V}_{j,n+1} = \frac{1}{n} [(n - j + 1)\bar{D} + (j - 1)\bar{C}] \quad (B2)$$

The position vectors of the points  $(1, k)$  and  $(n+1, k)$  correspondingly will be given by

$$\bar{V}_{1,k} = \frac{1}{n} [(n - k + 1)\bar{A} + (k - 1)\bar{D}] \quad (B3)$$

$$\bar{V}_{n+1,k} = \frac{1}{n} [(n - k + 1)\bar{B} + (k - 1)\bar{C}] \quad (B4)$$

If a line is drawn between  $\bar{V}_{j,1}$  and  $\bar{V}_{j,n+1}$  and another is drawn between  $\bar{V}_{1,k}$  and  $\bar{V}_{n+1,k}$ , their intersection will be  $\bar{V}_{j,k}$ , the position vector of the lower left corner of element  $jk$ . The line between  $\bar{V}_{j,1}$  and  $\bar{V}_{j,n+1}$  is divided by the intersection point in the same ratio as sides  $BC$  and  $AD$  are divided. Thus, in terms of  $\bar{V}_{j,1}$  and  $\bar{V}_{j,n+1}$ ,  $\bar{V}_{j,k}$  may be expressed as

## APPENDIX B – Continued

$$\bar{V}_{j,k} = \frac{1}{n} \left[ (n - k + 1) \bar{V}_{j,1} + (k - 1) \bar{V}_{j,n+1} \right] \quad (B5)$$

Substituting equations (B1) and (B2) into equation (B5) yields

$$\begin{aligned} \bar{V}_{j,k} = \frac{1}{n^2} & \left[ (n - j + 1)(n - k + 1) \bar{A} + (j - 1)(n - k + 1) \bar{B} \right. \\ & \left. + (j - 1)(k - 1) \bar{C} + (n - j + 1)(k - 1) \bar{D} \right] \end{aligned} \quad (B6)$$

This is the exact value of the position vector of the lower left corner of elemental area  $jk$  of the quadrilateral. The position vector of the centroid of the elemental quadrilateral may, except for extremely irregular shapes, be well approximated by

$$\begin{aligned} (\bar{V}_c)_{j,k} \approx \bar{V}_{j+\frac{1}{2},k+\frac{1}{2}} = \frac{1}{n^2} & \left[ \left( n - j + \frac{1}{2} \right) \left( n - k + \frac{1}{2} \right) \bar{A} + \left( j - \frac{1}{2} \right) \left( n - k + \frac{1}{2} \right) \bar{B} \right. \\ & \left. + \left( j - \frac{1}{2} \right) \left( k - \frac{1}{2} \right) \bar{C} + \left( n - j + \frac{1}{2} \right) \left( k - \frac{1}{2} \right) \bar{D} \right] \end{aligned} \quad (B7)$$

This is the mean of the position vectors of the four corners of the grid element and would also be the point of intersection if the grid were twice as fine.

The area of the element  $jk$  is found by dividing it into two triangles and taking half the magnitude of the cross product of two sides of each triangle. Since the upper right triangle of element  $jk$  is congruent to the lower left triangle of element  $j+1, k+1$ , the area of element  $jk$  is given by

$$\begin{aligned} (\Delta A)_{j,k} = \frac{1}{2} & \left| \left( \bar{V}_{j+1,k} - \bar{V}_{j,k} \right) \times \left( \bar{V}_{j,k+1} - \bar{V}_{j,k} \right) \right| \\ & + \frac{1}{2} \left| \left( \bar{V}_{j+2,k+1} - \bar{V}_{j+1,k+1} \right) \times \left( \bar{V}_{j+1,k+2} - \bar{V}_{j+1,k+1} \right) \right| \end{aligned} \quad (B8)$$

Substituting equation (B6) into equation (B8) yields

$$\begin{aligned} (\Delta A)_{j,k} = \frac{1}{2n^4} & \left| \left[ (n - k + 1)(\bar{B} - \bar{A}) + (k - 1)(\bar{C} - \bar{D}) \right] \times \left[ (n - j + 1)(\bar{D} - \bar{A}) + (j - 1)(\bar{C} - \bar{B}) \right] \right. \\ & \left. + \left[ (n - k)(\bar{B} - \bar{A}) + k(\bar{C} - \bar{D}) \right] \times \left[ (n - j)(\bar{D} - \bar{A}) + j(\bar{C} - \bar{B}) \right] \right| \end{aligned} \quad (B9)$$

## APPENDIX B – Concluded

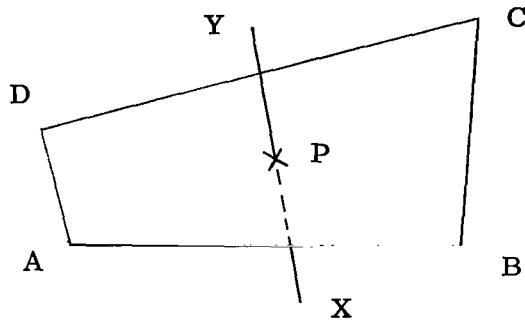
Expanding equation (B9) gives

$$\begin{aligned}
 (\Delta A)_{j,k} = \frac{1}{2n^4} \left\{ & \left[ (n - j + 1)(n - k + 1) + (n - j)(n - k) \right] \left| (\bar{B} - \bar{A}) \times (\bar{D} - \bar{A}) \right| \right. \\
 & + \left[ (j - 1)(n - k + 1) + j(n - k) \right] \left| (\bar{B} - \bar{A}) \times (\bar{C} - \bar{B}) \right| \\
 & + \left[ (n - j + 1)(k - 1) + (n - j)k \right] \left| (\bar{C} - \bar{D}) \times (\bar{D} - \bar{A}) \right| \\
 & \left. + \left[ (j - 1)(k - 1) + jk \right] \left| (\bar{C} - \bar{D}) \times (\bar{C} - \bar{B}) \right| \right\} \quad (B10)
 \end{aligned}$$

## APPENDIX C

### DETERMINATION OF SHADING

Five tests are made in the computer program to determine whether points  $X$  and  $Y$  are shaded from one another by the planar quadrilateral  $ABCD$ :



The first test determines whether the point of intersection  $P$  of the plane of  $ABCD$  by the line  $XY$  falls between points  $X$  and  $Y$ . Then for each of the four sides of  $ABCD$ , it is found whether  $P$  lies toward the inside or the outside of  $ABCD$  from that side.

The position vector of a point on the plane of the quadrilateral  $ABCD$  may be expressed as a linear combination of two vectors in the plane added to the position vector of a corner, say corner  $A$ :

$$\bar{r}_{\text{plane}} = \bar{A} + \alpha'(\bar{B} - \bar{A}) + \beta'(\bar{D} - \bar{A})$$

If  $\bar{P}$  is the position vector of the point of intersection of the plane by the line connecting  $X$  and  $Y$ , then

$$\bar{P} - \bar{X} = \gamma(\bar{Y} - \bar{X})$$

Setting  $\bar{P} = \bar{r}_{\text{plane}}$  gives

$$\alpha'(\bar{A} - \bar{B}) + \beta'(\bar{A} - \bar{D}) + \gamma(\bar{Y} - \bar{X}) = (\bar{A} - \bar{X})$$

or

$$\alpha' \bar{V}_1 + \beta' \bar{V}_2 + \gamma \bar{V}_3 = \bar{V}_4$$

This vector equation is, of course, three simultaneous linear equations in  $\alpha'$ ,  $\beta'$ , and  $\gamma$  — one for each vector component.

## APPENDIX C – Continued

The solution for  $\gamma$  will be carried out first, since the quadrilateral ABCD will be immediately eliminated as a shader if the condition  $0 < \gamma < 1$  is not met:

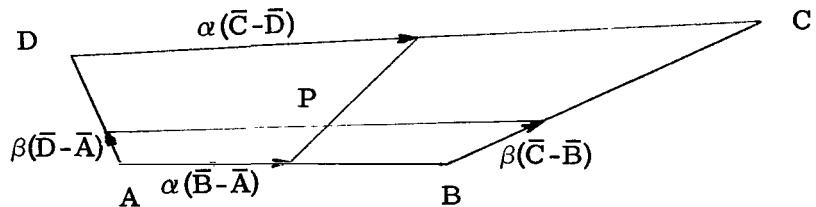
$$\gamma = \frac{\begin{vmatrix} (v_1)_x & (v_2)_x & (v_4)_x \\ (v_1)_y & (v_2)_y & (v_4)_y \\ (v_1)_z & (v_2)_z & (v_4)_z \end{vmatrix}}{\begin{vmatrix} (v_1)_x & (v_2)_x & (v_3)_x \\ (v_1)_y & (v_2)_y & (v_3)_y \\ (v_1)_z & (v_2)_z & (v_3)_z \end{vmatrix}} = \frac{\bar{v}_1 \cdot (\bar{v}_2 \times \bar{v}_4)}{\bar{v}_1 \cdot (\bar{v}_2 \times \bar{v}_3)}$$

If  $\alpha' < 0$ , the point lies to the left of side AD and if  $\beta' < 0$ , the point lies below side AB. These will also be evaluated one at a time:

$$\alpha' = \frac{\bar{v}_4 \cdot (\bar{v}_2 \times \bar{v}_3)}{\bar{v}_1 \cdot (\bar{v}_2 \times \bar{v}_3)} \quad \beta' = \frac{\bar{v}_1 \cdot (\bar{v}_4 \times \bar{v}_3)}{\bar{v}_1 \cdot (\bar{v}_2 \times \bar{v}_3)}$$

If  $\alpha'$  and  $\beta'$  are both greater than zero, more testing is required.

The reason for the primes is that  $\alpha'$  and  $\beta'$  are actually quadratics in the two linear parameters for the skewed coordinate frame formed by the quadrilateral. Let  $\alpha$  be a linear parameter characterizing a point moving from A to B or D to C as  $\alpha$  varies from 0 to 1. Let  $\beta$  be the parameter for points along AD or BC as shown in the following sketch:



## APPENDIX C – Continued

Then,

$$\begin{aligned}\bar{P} &= \bar{A} + \alpha(\bar{B} - \bar{A}) + \beta \left\{ \left[ \bar{D} + \alpha(\bar{C} - \bar{D}) \right] - \left[ \bar{A} + \alpha(\bar{B} - \bar{A}) \right] \right\} \\ &= \bar{A} + \alpha(\bar{B} - \bar{A}) + \beta(\bar{D} - \bar{A}) + \alpha\beta(\bar{A} - \bar{B} + \bar{C} - \bar{D})\end{aligned}$$

The vector  $(\bar{A} - \bar{B} + \bar{C} - \bar{D})$  lies in the plane of the quadrilateral and thus can be given as a linear combination of  $(\bar{B} - \bar{A})$  and  $(\bar{D} - \bar{A})$ :  $(\bar{A} - \bar{B} + \bar{C} - \bar{D}) = \lambda_1(\bar{B} - \bar{A}) + \lambda_2(\bar{D} - \bar{A})$ . The parameters  $\lambda_1$  and  $\lambda_2$  may be found by use of the vectors reciprocal to  $(\bar{B} - \bar{A})$  and  $(\bar{D} - \bar{A})$ , denoted by superscript  $R$  and characterized by

$$(\bar{B} - \bar{A})^R \cdot (\bar{B} - \bar{A}) = 1$$

$$(\bar{B} - \bar{A})^R \cdot (\bar{D} - \bar{A}) = 0$$

$$(\bar{D} - \bar{A})^R \cdot (\bar{B} - \bar{A}) = 0$$

$$(\bar{D} - \bar{A})^R \cdot (\bar{D} - \bar{A}) = 1$$

where, if  $\bar{N} = (\bar{B} - \bar{A}) \times (\bar{D} - \bar{A})$ , then

$$(\bar{B} - \bar{A})^R = \frac{(\bar{D} - \bar{A}) \times \bar{N}}{[(\bar{D} - \bar{A}) \times \bar{N}] \cdot (\bar{B} - \bar{A})}$$

$$(\bar{D} - \bar{A})^R = \frac{\bar{N} \times (\bar{B} - \bar{A})}{[\bar{N} \times (\bar{B} - \bar{A})] \cdot (\bar{D} - \bar{A})}$$

Now,

$$(\bar{B} - \bar{A})^R \cdot (\bar{A} - \bar{B} + \bar{C} - \bar{D}) = \lambda_1$$

and

$$(\bar{D} - \bar{A})^R \cdot (\bar{A} - \bar{B} + \bar{C} - \bar{D}) = \lambda_2$$

Thus,

$$\bar{P} = \bar{A} + \alpha(\bar{B} - \bar{A}) + \beta(\bar{D} - \bar{A}) + \alpha\beta[\lambda_1(\bar{B} - \bar{A}) + \lambda_2(\bar{D} - \bar{A})]$$

## APPENDIX C – Continued

or

$$\bar{P} = \bar{A} + \alpha(1 + \lambda_1\beta)(\bar{B} - \bar{A}) + \beta(1 + \lambda_2\alpha)(\bar{D} - \bar{A})$$

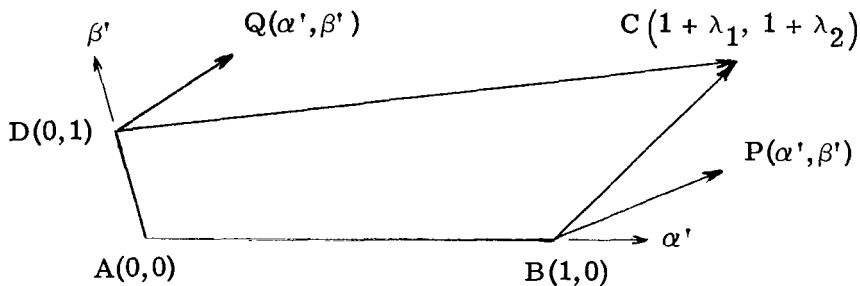
and

$$\alpha' = (1 + \lambda_1\beta)\alpha$$

$$\beta' = (1 + \lambda_2\alpha)\beta$$

from which the values of  $\alpha$  and  $\beta$  can be found directly.

An advantageous alternative to finding  $\alpha$  and  $\beta$  is found by plotting the quadrilateral ABCD in  $\alpha', \beta'$  coordinates:



The  $\alpha'$  and  $\beta'$  coordinates of the corners are found from the two equations for  $\alpha'$  and  $\beta'$  in terms of  $\alpha$  and  $\beta$ , with  $\alpha$  and  $\beta$  having a value of 0 or 1 at the corners.

The points P and Q represent possible points of intersection between the plane of ABCD and a line connecting two points of interest. If point P lies on the right-hand side of line BC, then the single nonzero component of the cross product of vector  $(\bar{P} - \bar{B})$  upon  $(\bar{C} - \bar{B})$  will be positive. Thus for no shading, this third component will be given by

$$[(\bar{P} - \bar{B}) \times (\bar{C} - \bar{B})]_3 = (\alpha' - 1)(1 + \lambda_2) - \beta'\lambda_1 > 0$$

If this quantity is zero or negative,  $\bar{P}$  lies on or to the left of line BC. Similarly, the condition that the point lie above line CD is

$$[(\bar{C} - \bar{D}) \times (\bar{Q} - \bar{D})]_3 = (1 + \lambda_1)(\beta' - 1) - \lambda_2\alpha' > 0$$

## APPENDIX C – Concluded

The five tests for shading, in the order of execution, are thus

$$(1) \quad 0 < \gamma < 1$$

$$(2) \quad \alpha' \geq 0$$

$$(3) \quad \beta \geq 0$$

$$(4) \quad (\alpha' - 1)(1 + \lambda_2) - \beta' \cdot \lambda_1 \leq 0$$

$$(5) \quad (\beta' - 1)(1 + \lambda_1) - \alpha' \cdot \lambda_2 \leq 0$$

If any one of these fails, there is no shading between the two points by the quadrilateral ABCD. They must all hold for shading to occur.

## REFERENCES

1. Eckert, E. R. G.; and Drake, Robert M., Jr.: Heat and Mass Transfer. McGraw-Hill Book Co., Inc., 1959.
2. Goerss, R. H.: Computer Analysis of Satellite Thermal Behavior. Spacecraft and Space Systems, Radio Corp. Amer., 1966, pp. 30-33.
3. Gebhart, Benjamin: Heat Transfer. McGraw-Hill Book Co., Inc., c.1961.

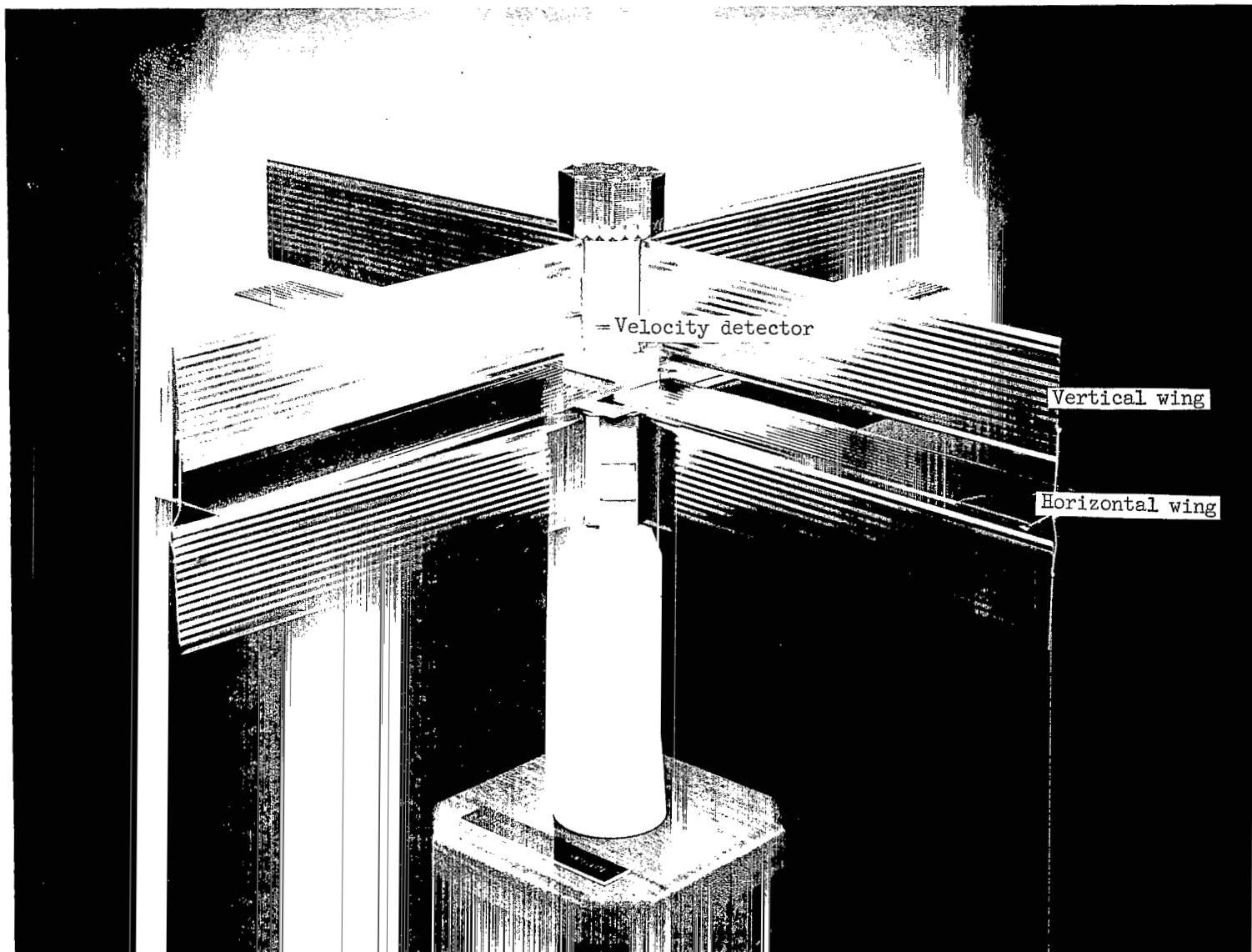


Figure 1.- Location of three sets of sample nodes on the Meteoroid Technology Satellite.

L-69-657

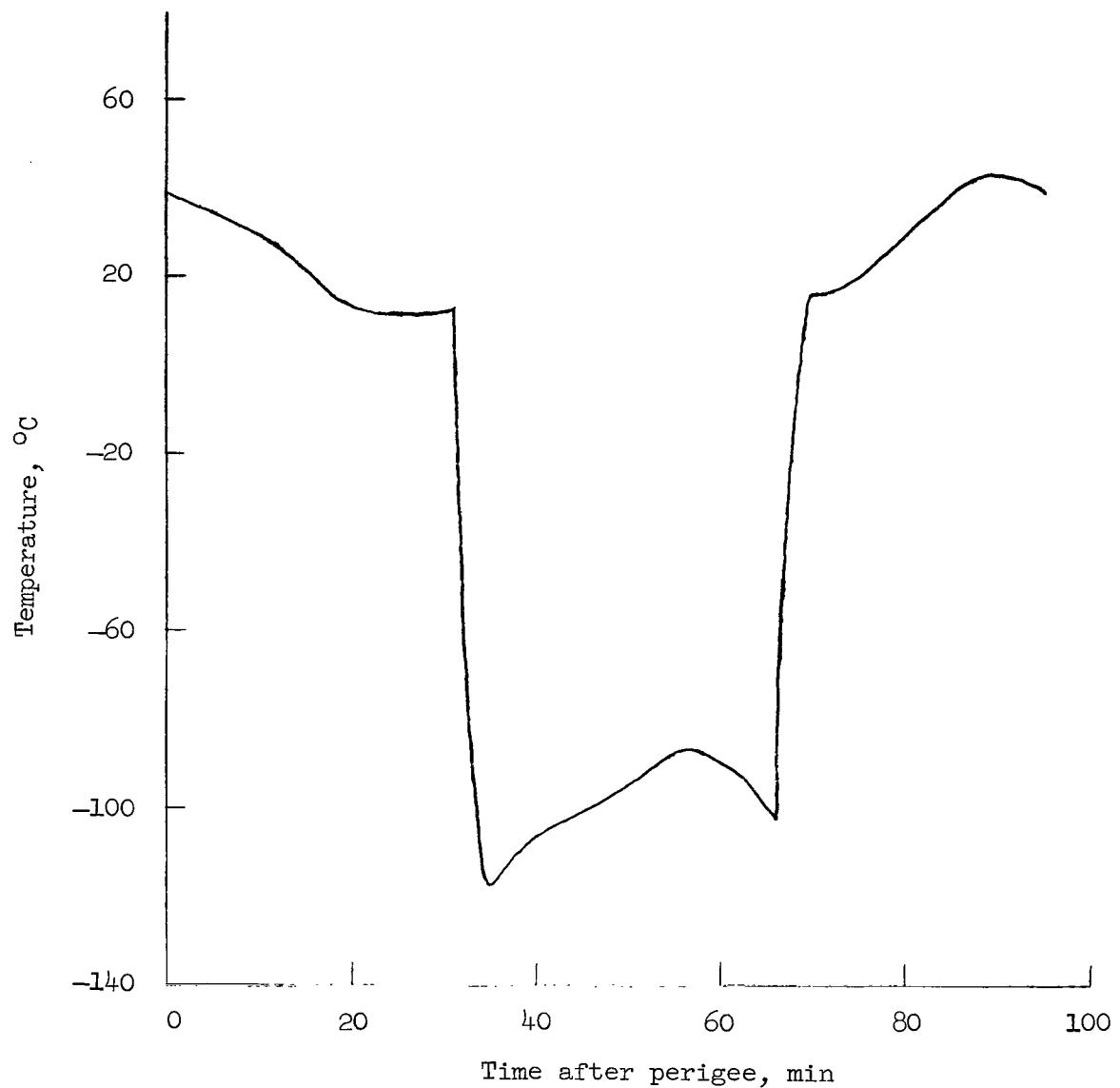


Figure 2.- Temperature history of outer face of upper velocity detector.

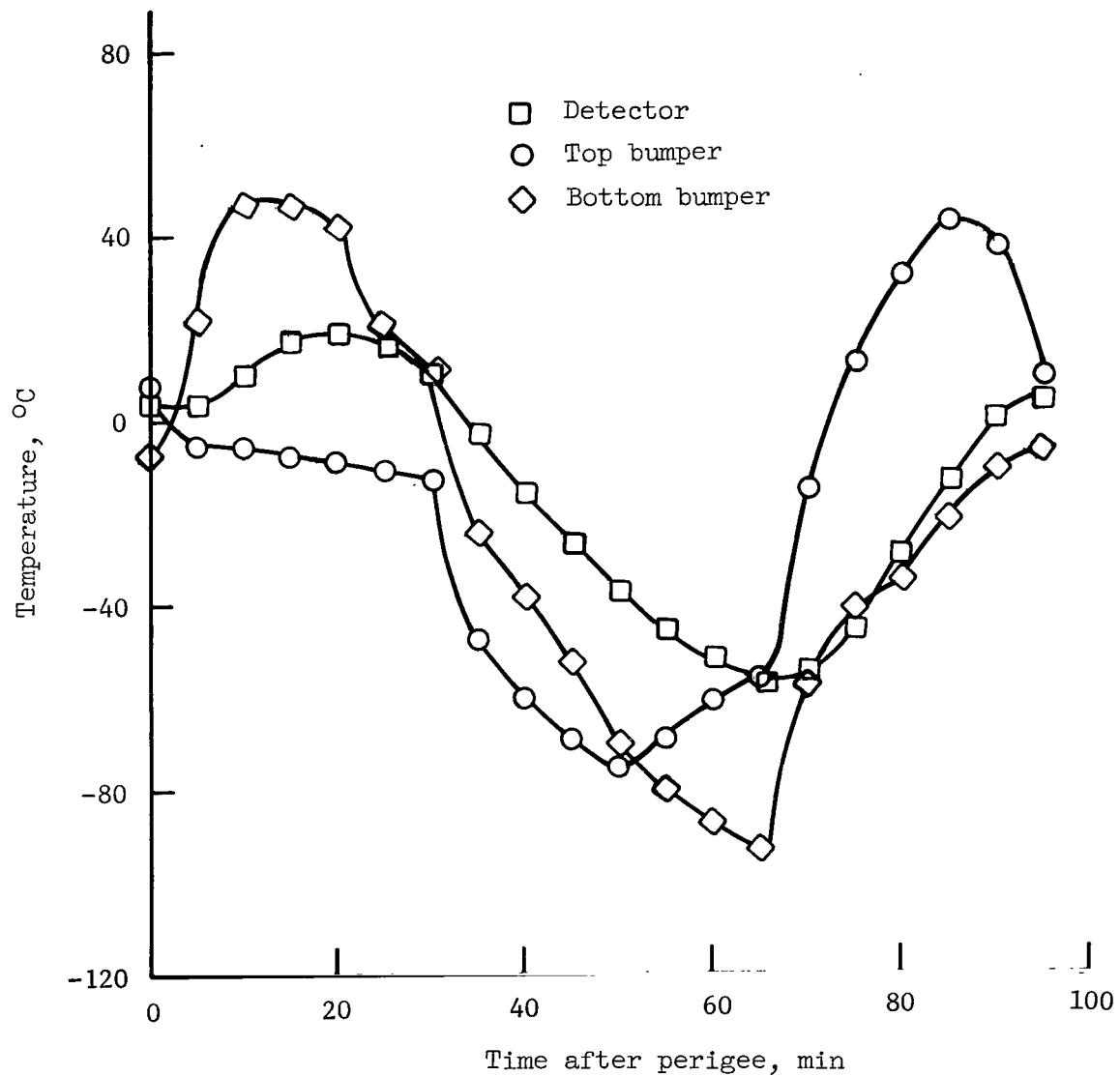


Figure 3.- Temperature histories of horizontal wing detector and its bumper shields.

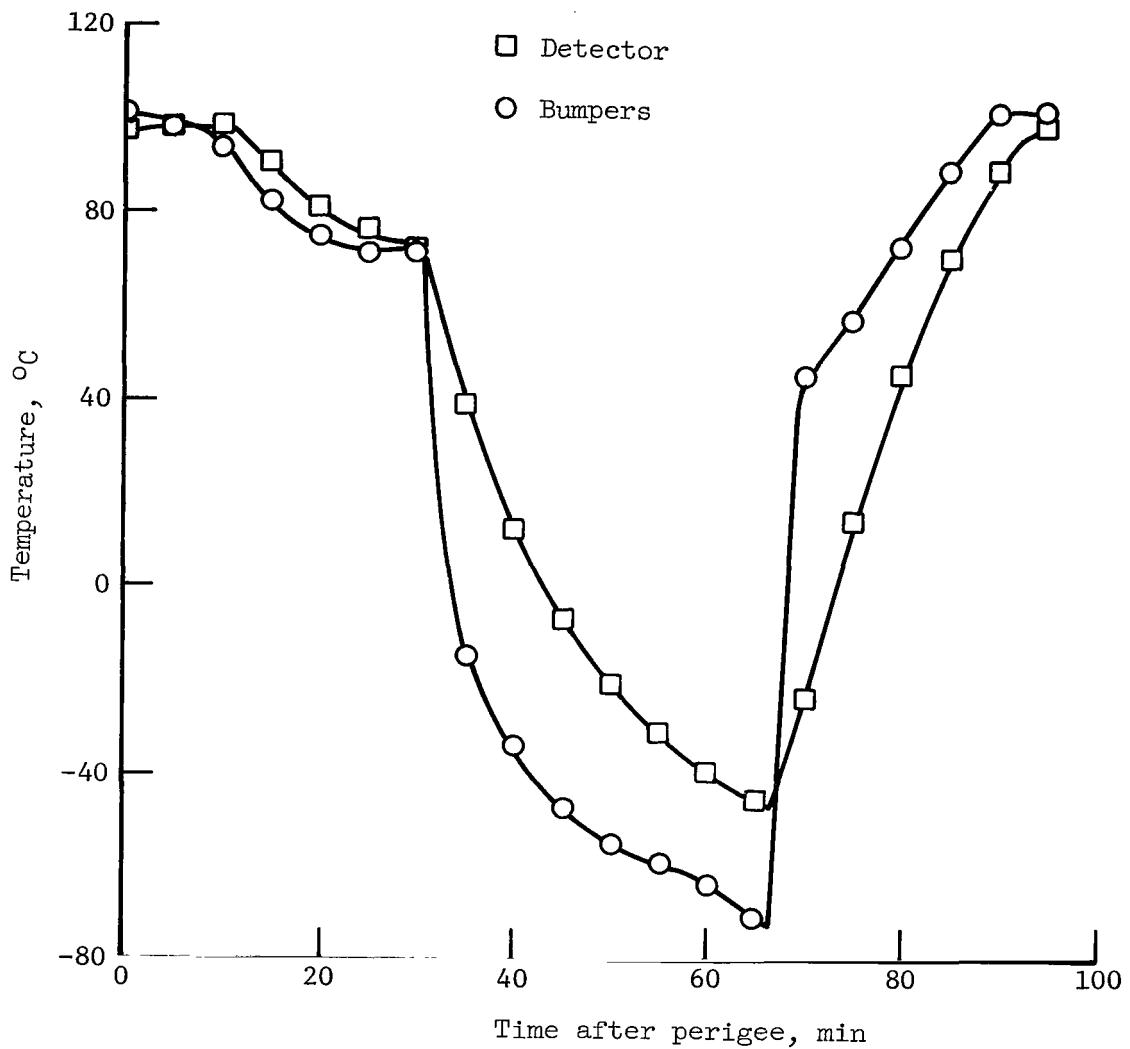


Figure 4.- Temperature history of vertical wing detector and its bumper shields.

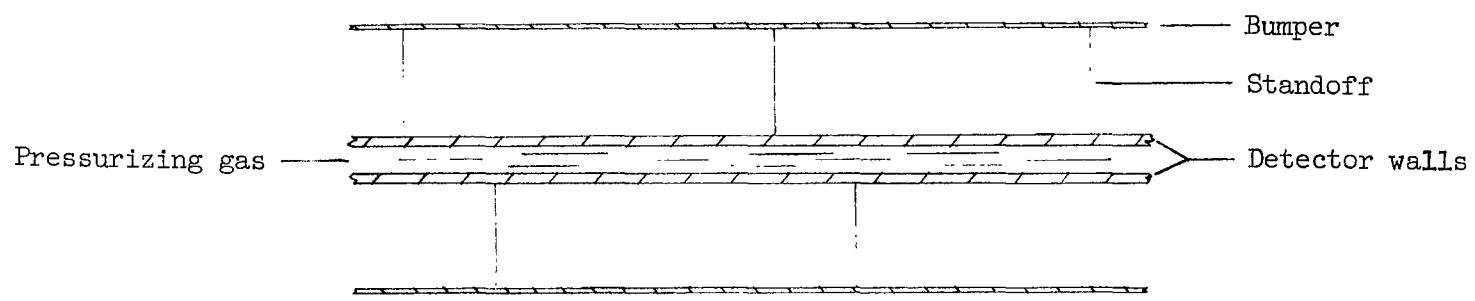


Figure 5.- Schematic cross section of a pressurized-cell meteoroid detector with bumpers.

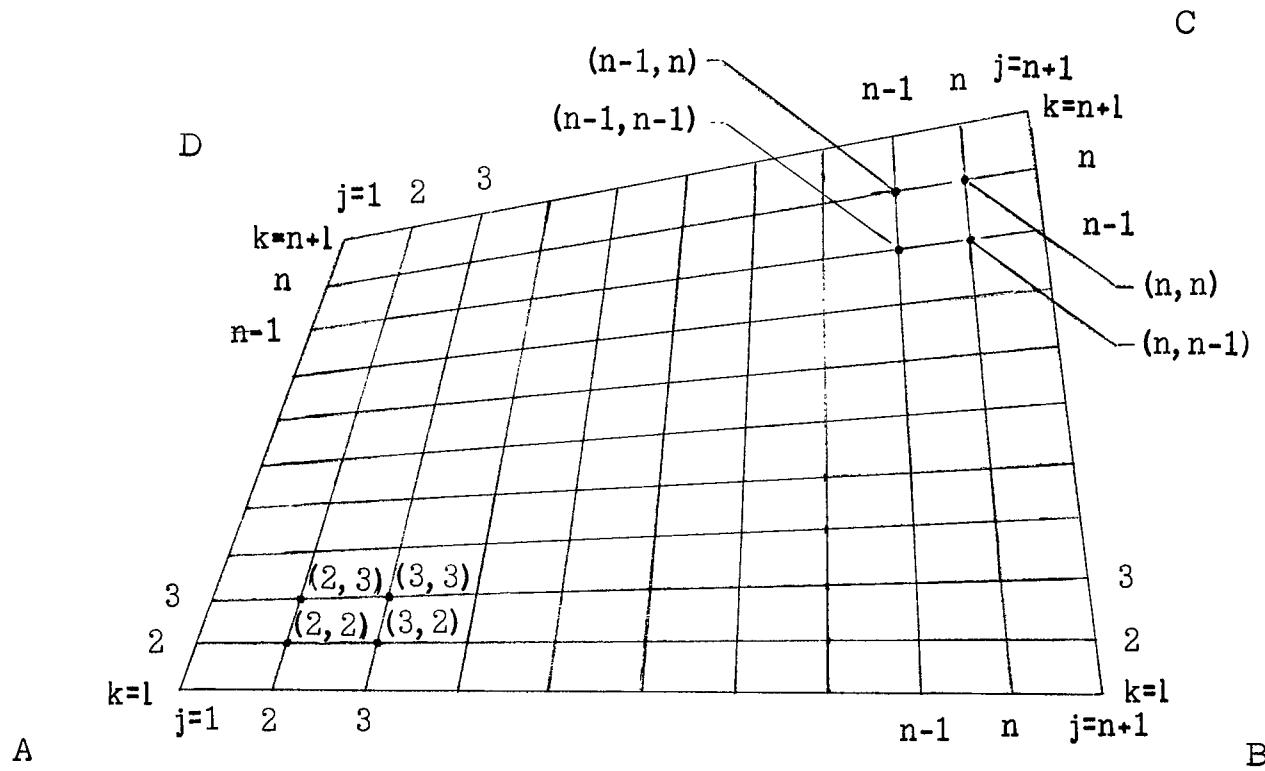
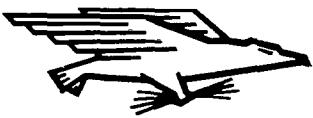


Figure 6.- Plane quadrilateral with  $n \times n$  grid formed by dividing each side into  $n$  equal segments, with coordinate parameters  $j$  and  $k$  illustrated.



NATIONAL AERONAUTICS AND SPACE ADMINISTRATION  
WASHINGTON, D. C. 20546  
OFFICIAL BUSINESS

FIRST CLASS MAIL



POSTAGE AND FEES PAID  
NATIONAL AERONAUTICS AND  
SPACE ADMINISTRATION

05U 001 58 51 3DS 70195 00903  
AIR FORCE WEAPONS LABORATORY /WLOL/  
KIRTLAND AFB, NEW MEXICO 87117

ATT E. LOU BOWMAN, CHIEF, TECH. LIBRARY

POSTMASTER: If Undeliverable (Section 158  
Postal Manual) Do Not Return

*"The aeronautical and space activities of the United States shall be conducted so as to contribute . . . to the expansion of human knowledge of phenomena in the atmosphere and space. The Administration shall provide for the widest practicable and appropriate dissemination of information concerning its activities and the results thereof."*

— NATIONAL AERONAUTICS AND SPACE ACT OF 1958

## NASA SCIENTIFIC AND TECHNICAL PUBLICATIONS

**TECHNICAL REPORTS:** Scientific and technical information considered important, complete, and a lasting contribution to existing knowledge.

**TECHNICAL NOTES:** Information less broad in scope but nevertheless of importance as a contribution to existing knowledge.

**TECHNICAL MEMORANDUMS:** Information receiving limited distribution because of preliminary data, security classification, or other reasons.

**CONTRACTOR REPORTS:** Scientific and technical information generated under a NASA contract or grant and considered an important contribution to existing knowledge.

**TECHNICAL TRANSLATIONS:** Information published in a foreign language considered to merit NASA distribution in English.

**SPECIAL PUBLICATIONS:** Information derived from or of value to NASA activities. Publications include conference proceedings, monographs, data compilations, handbooks, sourcebooks, and special bibliographies.

**TECHNOLOGY UTILIZATION PUBLICATIONS:** Information on technology used by NASA that may be of particular interest in commercial and other non-aerospace applications. Publications include Tech Briefs, Technology Utilization Reports and Notes, and Technology Surveys.

*Details on the availability of these publications may be obtained from:*

SCIENTIFIC AND TECHNICAL INFORMATION DIVISION  
NATIONAL AERONAUTICS AND SPACE ADMINISTRATION  
Washington, D.C. 20546